

*Article*

# Ornithopter Type Flapping Wings for Autonomous Micro Air Vehicles

Sutthiphong Srigrarom <sup>1,\*</sup> and Woei-Leong Chan <sup>2</sup>

<sup>1</sup> Aerospace Systems, University of Glasgow Singapore, 500, Dover Rd., #T1A-02-24, Singapore 139651

<sup>2</sup> Temasek Laboratories, National University of Singapore, #09-02, 5A Engineering Drive 1, Singapore 117411; E-Mail: tslcwl@nus.edu.sg

\* Author to whom correspondence should be addressed; E-Mail: spot.srigrarom@glasgow.ac.uk; Tel.: +65-6908-6033.

Academic Editors: David Anderson and Rafic Ajaj

*Received: 2 February 2015 / Accepted: 4 May 2015 / Published: 13 May 2015*

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**Abstract:** In this paper, an ornithopter prototype that mimics the flapping motion of bird flight is developed, and the lift and thrust generation characteristics of different wing designs are evaluated. This project focused on the spar arrangement and material used for the wings that could achieve improved performance. Various lift and thrust measurement techniques are explored and evaluated. Various wings of insects and birds were evaluated to understand how these natural flyers with flapping wings are able to produce sufficient lift to fly. The differences in the flapping aerodynamics were also detailed. Experiments on different wing designs and materials were conducted and a paramount wing was built for a test flight. The first prototype has a length of 46.5 cm, wing span of 88 cm, and weighs 161 g. A mechanism which produced a flapping motion was fabricated and designed to create flapping flight. The flapping flight was produced by using a single motor and a flexible and light wing structure. A force balance made of load cell was then designed to measure the thrust and lift force of the ornithopter. Three sets of wings varying flexibility were fabricated, therefore lift and thrust measurements were acquired from each different set of wings. The lift will be measured in ten cycles computing the average lift and frequency in three different speeds or frequencies (slow, medium and fast). The thrust measurement was measured likewise but in two cycles only. Several observations were made regarding the behavior of flexible flapping wings that should aid in the design of future flexible flapping wing vehicles. The wings angle or phase characteristic were analyzed too and studied.

The final ornithopter prototype weighs only 160 g, has a wing span of 88.5 cm, that could flap at a maximum flapping frequency of 3.869 Hz, and produce a maximum thrust and lift of about 0.719 and 0.264 N respectively. Next, we proposed resonance type flapping wing utilizes the near resonance phenomenon of a two-degree of freedom elastic system, that is, the wing is supported by the springs for flapping and feathering motions. Being oscillated close to the resonance frequency of the system, only by the torque in flapping motion, the amplitude gained is a few times higher than that of normal case. The first prototype was made from acrylic using a laser cutting machine. The wings were made up of carbon rods and kite material Ripstop. First test showed that the wings were too heavy for the mechanism to work. The third prototype was a smaller single gear crank design which was fabricated using a 3D printer. Initial test proved that the second prototype could withstand the high frequency flapping and near resonance amplitude as designed. With remote control, the third prototype was able to take off, climb, cruise and land in flapping mode successfully.

**Keywords:** ornithopter-like flapping wing; albatross-like wing; micro air vehicle (MAV)

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## 1. Introduction

There is a growing recognized need for miniature flight vehicles with multifunctional capabilities, such as micro air vehicles (MAVs) for both military and civilian surveillance [1–4]. Flapping wing flight of birds provides us with a sophisticated example of utilizing unsteady aerodynamics to mechanize the miniature flight structures at low Reynolds numbers ( $10^3$ – $10^5$ ) [5,6]. We attempt to mimic both the long-distance birds, due to their natural long-endurance manner, and their high lift production during take-off (start-up). The albatross, as shown in Figures 1 and 2, is chosen to represent long distance migratory birds. It is about the size of our intended flapping wing based surveillance MAV. We hope to mimic albatross's flapping flight to achieve this long-distance characteristic. It is used for investigating flow characteristic aiming at better design of flapping MAV. The wingspan of the albatross is 60 cm. The flapping pattern of the albatross is of the avian type, *i.e.*, vertical motion as shown in Figure 3. Our model simulated the complete, three-dimensional, unsteady flow fields around this type of wing with large-scale vortices.



**Figure 1.** Albatross in flight.

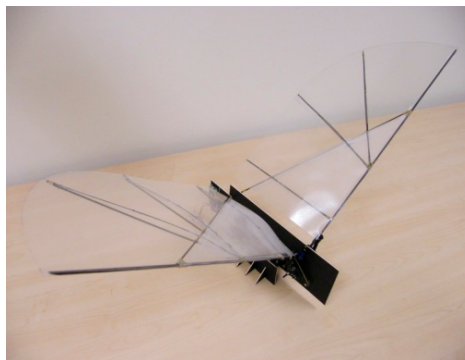


**Figure 2.** Albatross wing model.

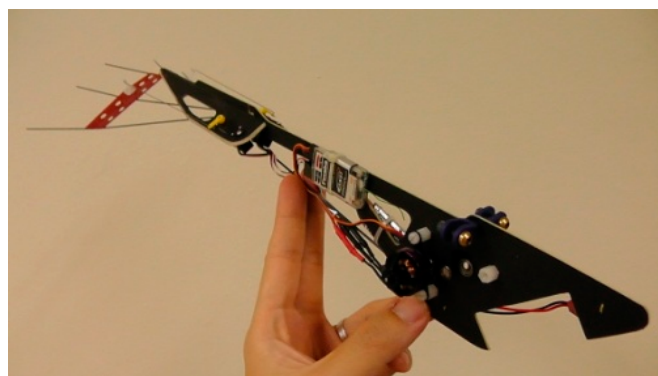


**Figure 3.** The flapping pattern of albatross is avian type, *i.e.*, more up/down vertically.

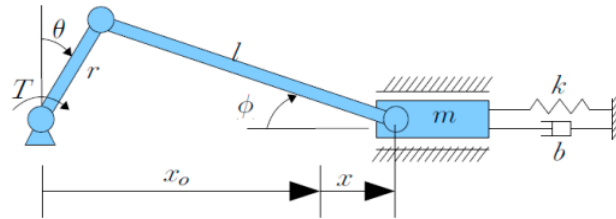
At University of Glasgow Singapore, we are developing the resonance type flapping wing models, as shown in Figures 4 and 5 below. The proposed resonance type flapping wing will utilize the resonance phenomenon of a two-degree of freedom elastic system, that is, the wing is supported by the springs for flapping and feathering motions, being oscillated, at the resonance frequency of the system, as shown in Figure 6 [7]. The amplitudes of flapping and feathering motions and the phase angle between them are controlled by changing the amount of the damping.



**Figure 4.** Near-resonance ornithopter-like flapping wing model.



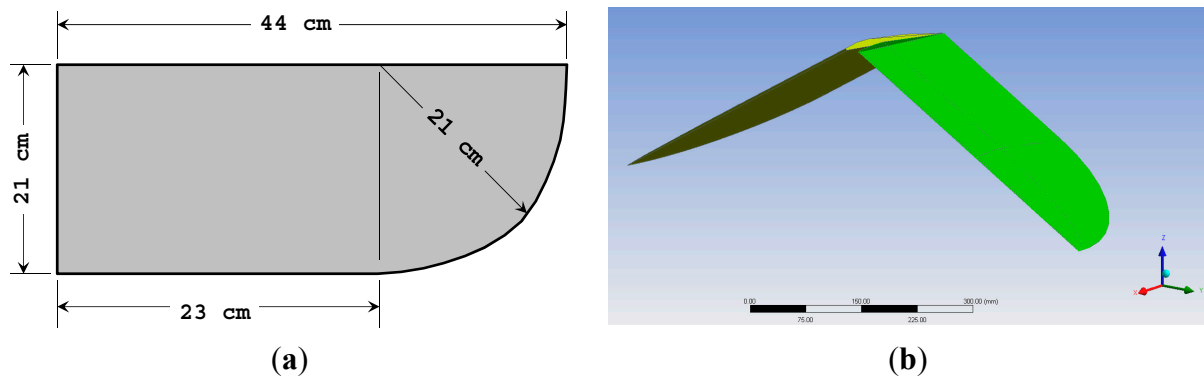
**Figure 5.** Near-resonance ornithopter-like flapping wing mechanism.



**Figure 6.** Concept of resonance-type flapping mechanism [7].

## 2. Computational Fluid Dynamics

Firstly, we examined the flight dynamics as well as unsteady flow characteristics of the flapping wings for long-distance bird (e.g., albatross-like) by Computational Fluid Dynamics (CFD). These birds mostly flap their wings about their body axis (chordwise) with little change in twist (spanwise), as depicted in Figure 7 [8]. Therefore, for preliminary analyses, we apply only chordwise flapping to our model.

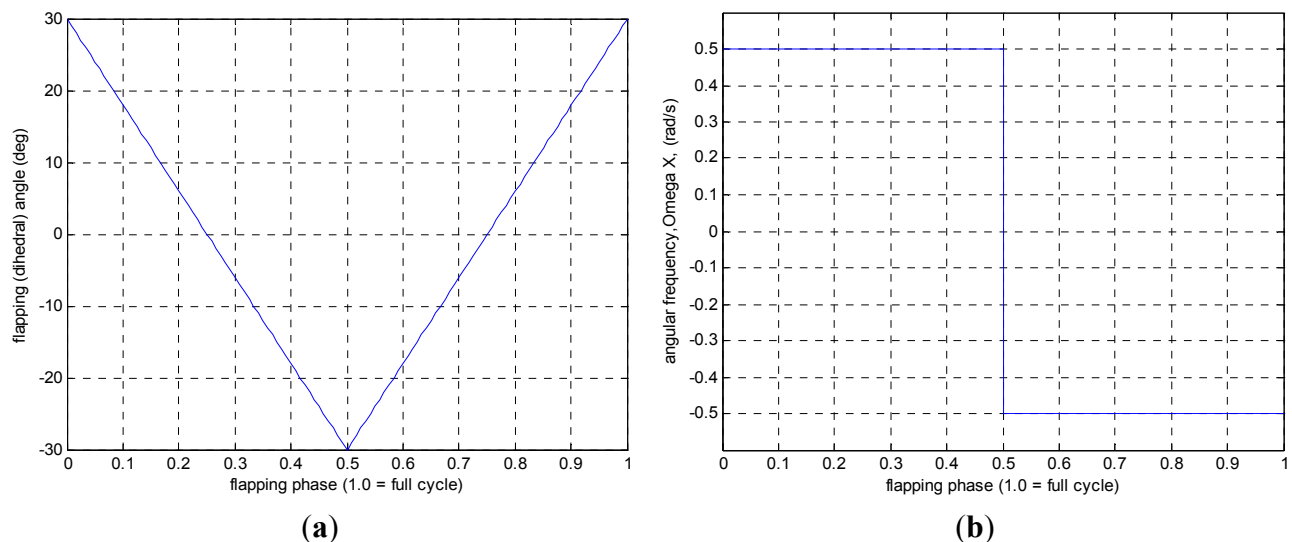


**Figure 7.** (a) Simplified albatross flapping model with detailed geometry; (b) The wing is shown at  $-15^\circ$  flapping position about its body ( $X$ ) axis.

The flow around the flapping wing was simulated by using ANSYS Fluent<sup>®</sup> (ANSYS Inc., Canonsburg, PA, USA) unsteady three-dimensional compressible Navier–Stokes equations. The machine that ran these problems was a 64 bit computer, Intel<sup>®</sup> Core i7-2600 CPU at 3.4 GHz. It had eight processors at 16 GByte RAM. The geometric model of the albatross wing was the idealization of an albatross. The model wing had a wing span of 30 cm, mean wing chord length of 5 cm, a thickness of 2.5% of the mean wing chord length, as shown in Figure 7. The computing domain extended to 50-chord lengths in all directions around the full model wing, and that, there were about  $10^7$  meshes of the tetrahedral type. The flow condition was unsteady transient flow, with the built-in Large–Eddy Simulation (LES) turbulence model.

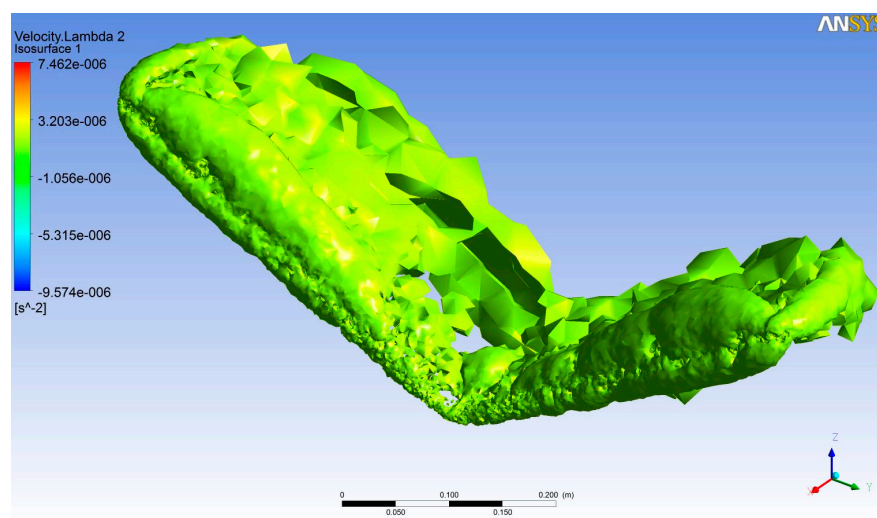
The chosen wing's geometry is shown in Figure 7 below, of which Wingspan = 0.88, Wing Surface Area =  $1.66 \times 10^{-1} \text{ m}^2$ . This wing was originally set at rest. From past studies [8–11], the starting flapping frequency of the albatross wing was at 0.5 rad/s, corresponding to the reduced frequency  $k$  (defined as  $\omega c/2U_\infty$ ) of 0.0025. The take-off speed of the albatross (*i.e.*, the freestream velocity in body-fitted coordinates),  $U_\infty$  was approximately 5 m/s. This gave the Reynolds number, based on chord  $Re$  of 25,000. The wing was set to flap down and up about its body axis (chordwise), mimicking

the start-up of the flapping motion (taking-off) and return to its initial position. The flapping (dihedral) angle changed from  $+30^\circ$  to  $-30^\circ$  in downstroke motion, and from  $-30^\circ$  to  $+30^\circ$  in upstroke motion. The input angular velocity about its chord ( $\omega_x$ ) is shown in Figure 8. This downwards stroke in the first half of the cycle is the lift generation stroke, whereas the upwards stroke in the latter half of the cycle is the recovery stroke.

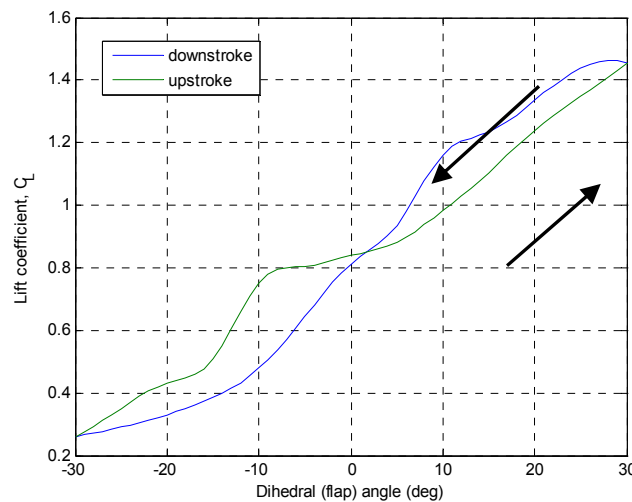


**Figure 8.** (a) Flapping patterns for simplified albatross flapping model about its body axis (chordwise,  $\omega_x$ ); and (b) the corresponding angular velocity, in each flapping cycle.

Initially, the wing was positioned at  $+30$  degree dihedral angle. When the flapping motion started, the wing flapped downwards about its chord. At the outer half of the wing the flow separated from the wing, forming the start-up vortex which more visible at the top. There was flow separation at the trailing edge, closer to the root, at the area near the scapula (inboard of the wing). This leads to the creation of strong leading edge vortex [8] as shown in Figure 9, resulting in high lift (Figure 10) at the beginning of this downstroke motion (at  $+30$  degree dihedral angle).



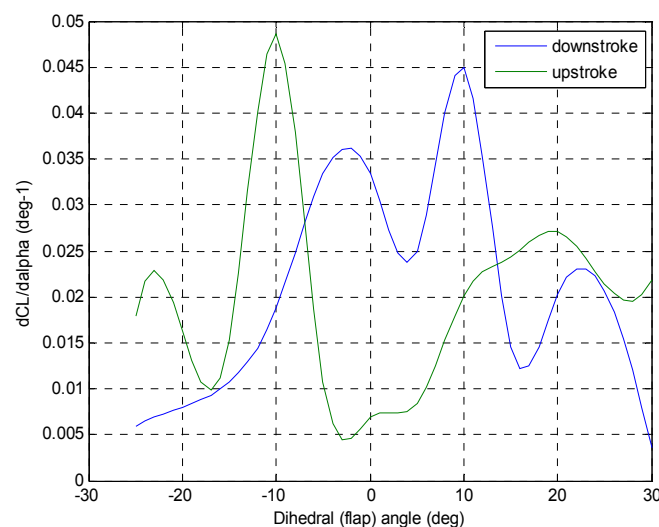
**Figure 9.** Vortex structure on the wing, shortly after start-up. The strong leading edge vortex on the upper part is clearly shown.



**Figure 10.** Lift coefficient during downstroke and upstroke motion of the wing.

As time goes on the flow separates across the wing, forming the strong leading edge vortex over the wing. In the inboard area near the trailing edge and scapula, the wake (shear layers) becomes more obvious. The lift is generated more by to this push-down motion of the wing. After mid-point (dihedral angle =  $0^\circ$ ), the start-up vortex becomes bigger, and the flow separates from the wing entirely. At the leading edge near the alula (about half way between the root and the tip). The wake at the trailing edge of the wing also breaks up into outer and inner parts.

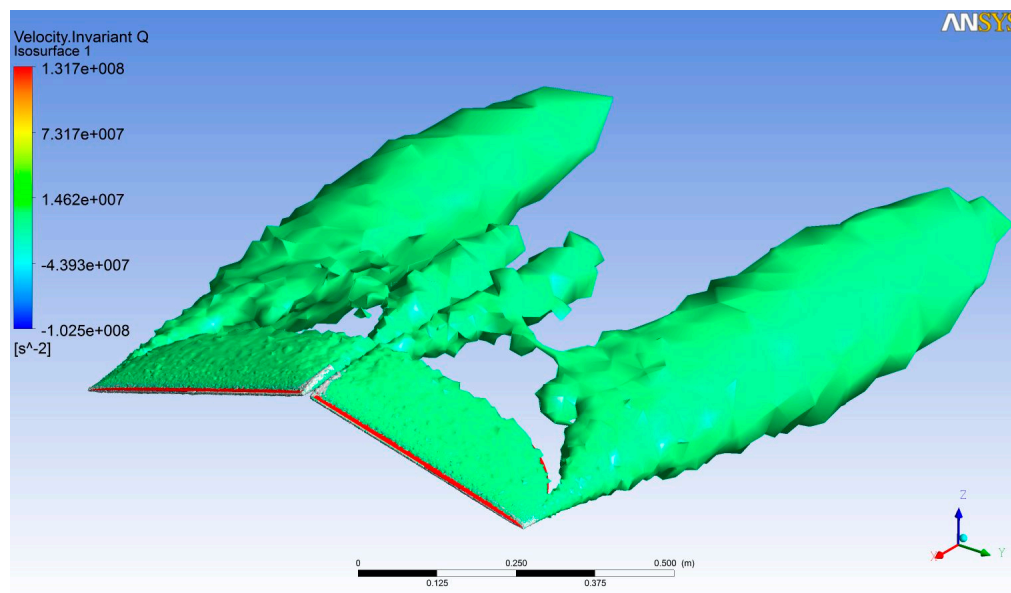
Figure 10 shows the lift coefficient in both downstroke and upstroke motion of the wing. The  $C_L$  values are obtained from directly from ANSYS Fluent<sup>®</sup> software. The locus of the lift coefficient in both downstroke and upstroke motion resembles the inclined Figure 8, as reported in many other literatures (e.g., [4,6,8,10,12–15]). Figure 11 shows the lift coefficients derivatives with respect to flapping angle ( $dC_L/d\alpha$ ) this is for flight dynamics and control consideration.



**Figure 11.** The lift coefficients derivatives with respect to flapping angle ( $dC_L/d\alpha$ ), from  $C_L$ .

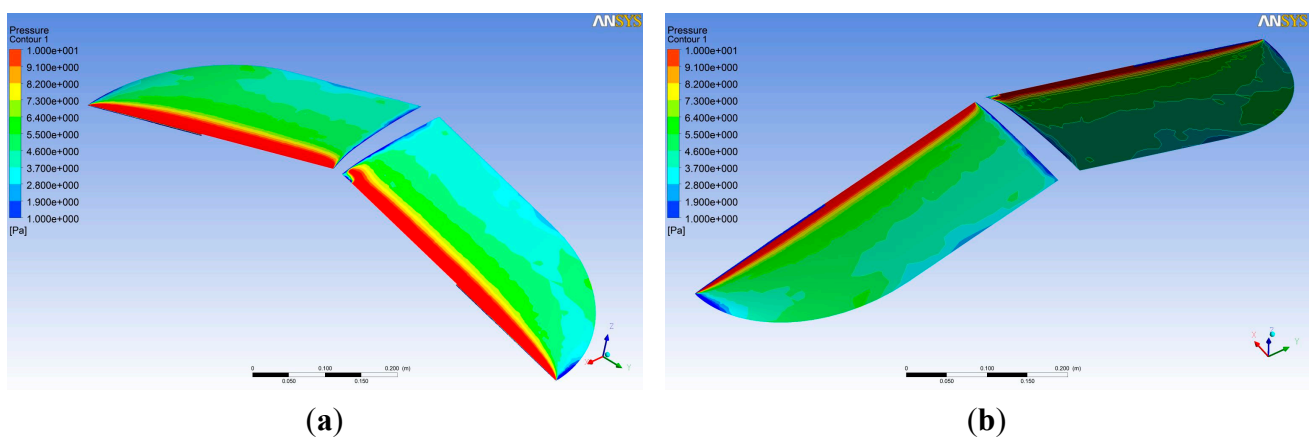
Figure 12 shows the vortex structure (invariant-Q) at dihedral angle =  $-15^\circ$ . The leading edge vortex is dominant, covering the entire the wing. At the wing tip, the wingtip vortices is clear together with the wake behind the wing.





**Figure 12.** Isosurface of the vorticity on the upper part of the albatross wing, at dihedral angle =  $-15^\circ$ . The leading edge vortex covers the whole wing. The trailing edge vortices on both ends are outstanding.

Figure 13 shows the surface pressure on the albatross wing also at dihedral angle =  $-15^\circ$ . At the tip, there is region of lower pressure, shown in blue color. This is the area where the force is less generated. So, we deduce the albatross does maneuvering by flapping its entire wingspan at different amplitudes than using (flapping or twisting) its wing tip only.



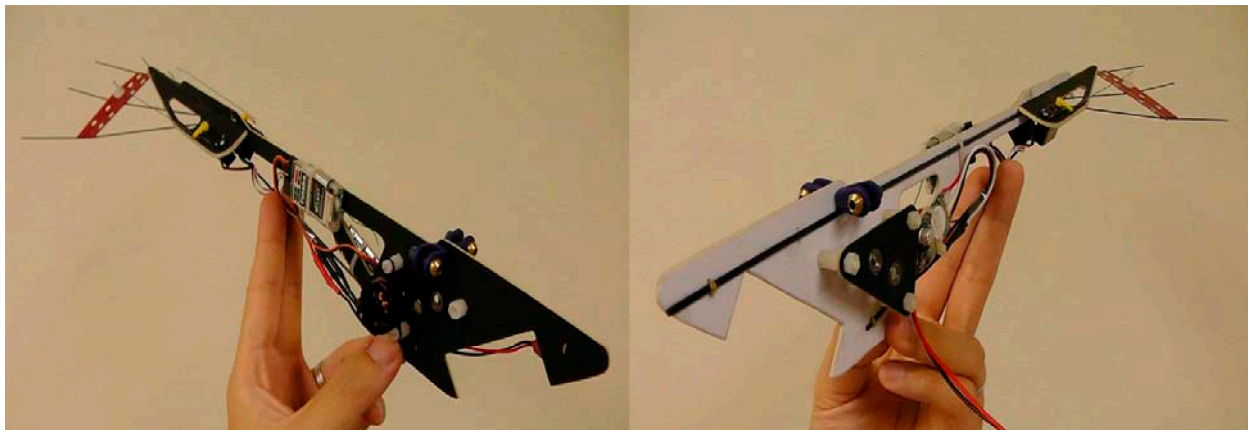
**Figure 13.** Surface pressure on the albatross wing at dihedral angle =  $-15^\circ$ , upper surface (a) and lower surface (b).

These unsteady flow results show that the albatross generates lift on its wing mainly by vortex lift mechanism, as reported in other literatures ([15–28]). The lift is generated by flapping its entire left and right wings at different amplitudes than using (flapping or twisting) its wing tip only. Any additional spanwise flapping (pitching) will fine-tune its position in flight. Therefore, for the sake of developing flapping-wing like MAV (next section), we could introduce only the chordwise flapping motion and neglect the spanwise flapping motion.

### 3. UGS Flapping Wing MAV Prototypes 1

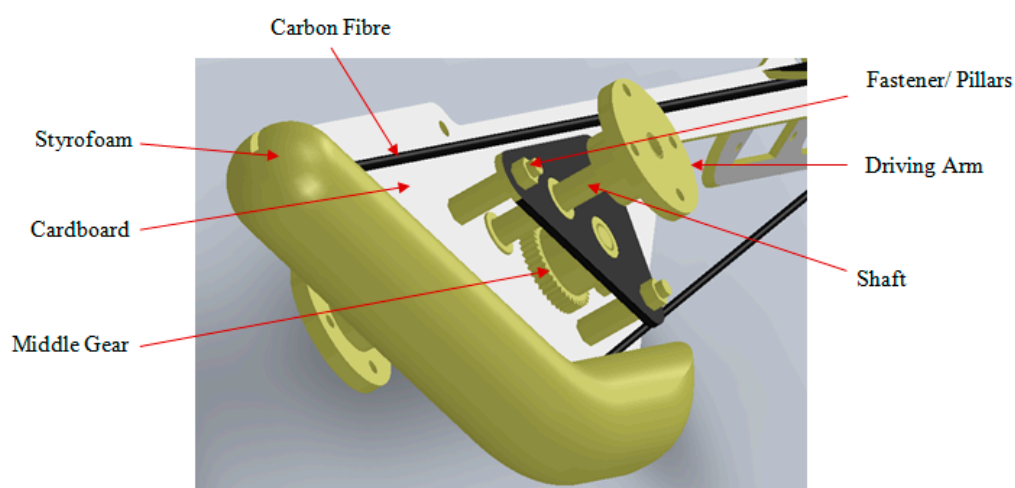
#### 3.1. Mechanism

A flapping wing mechanism capable of producing flapping motion is designed and built. An overall view of this mechanism with a length of 46.5 cm can be seen in Figure 14. This figure also depicts the relative scale between the flapping mechanism and the tail. This mechanism is powered by a small brushless motor and has a maximum output of 55 W that weight only 15 g. A Lithium–Polymer battery that has a nominal voltage of 7.4 V is required to drive the motor.



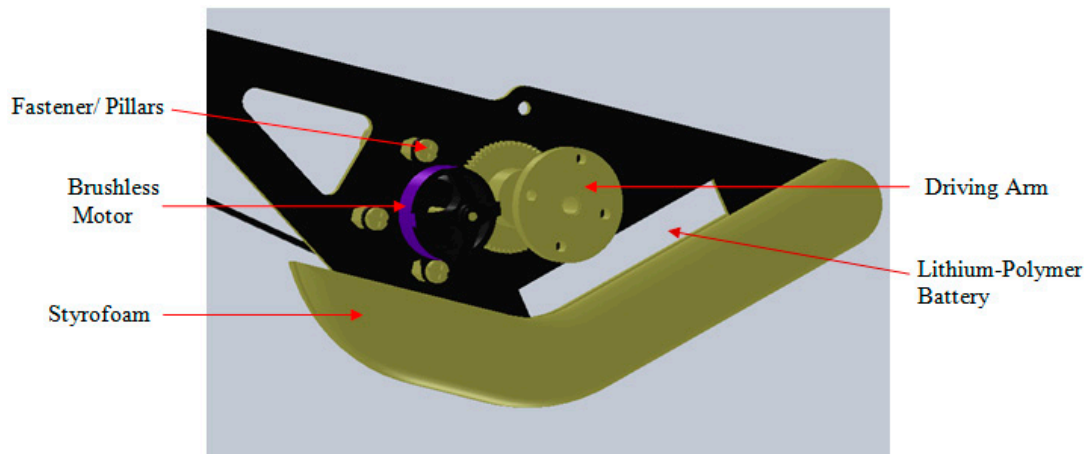
**Figure 14.** Overall view of entire flapping mechanism.

A complete set of computer aided design (CAD) models are developed to ensure an accurate fit between parts. The software used to generate CAD models is done using CATIA (Computer Aided Three-dimensional Interactive Application (Dassault Systèmes, Cedex, France)). The various parts of the mechanism can be seen in Figures 15 and 16.



**Figure 15.** Left side view of flapping mechanism (computer aided design (CAD) model).





**Figure 16.** Right side view of flapping mechanism (CAD model).

From Figure 16 the brushless DC motor is mounted onto the fuselage of the ornithopter using three nylon pillars. The motor is powered directly by Lithium–Polymer battery. In order for both wings to flap in sync, a single shaft is installed and attached onto both sides of the driving arms. The driving arms are then attached along with the wings. The Styrofoam at the front side of the fuselage is used to reduce the impact if the ornithopter would to fall. Carbon fiber sticks are used to strengthen the fuselage to prevent twisting and bending stress when it flaps. The Lithium–Polymer battery is placed in the front fuselage, so that the center of gravity for the prototype is achievable.

It is important to make the mechanism as light as possible, therefore the lightest, and most cost effective materials available are considered. Initially, carbon fiber which is strong and light-weight is one of the considerations. However, there are limited resources to manufacture and cut into the desired shape. As a result, the next option is to use cardboard, which is equivalent light-weight and easy to cut but not as durable as carbon fiber. Each component and sub-assembly for this ornithopter is weighted as shown in Table 1.

**Table 1.** Weight of each Component and Sub-assembly.

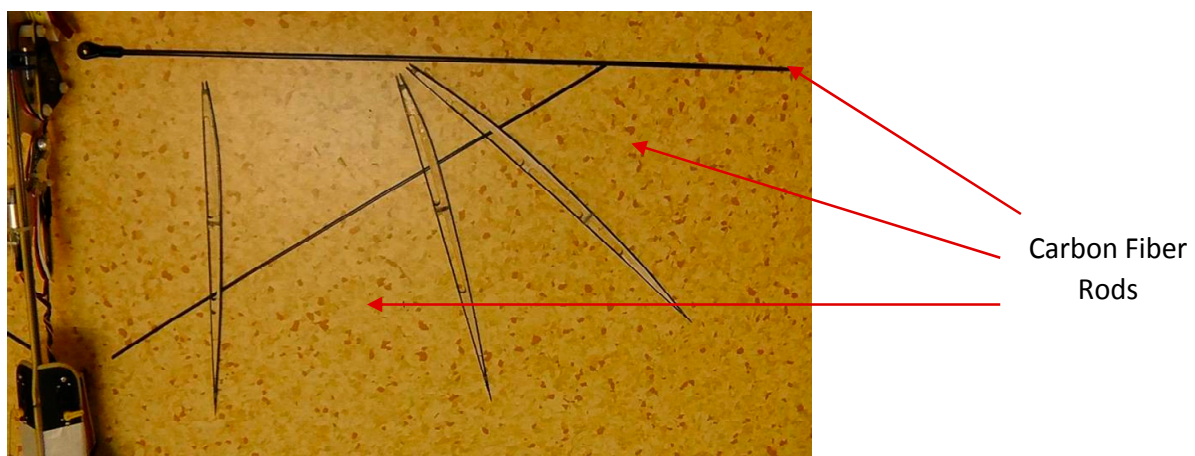
Component and Sub-assembly	Weight (g)
Kypom Lithium–Polymer Battery	29
Hacker A10-A12s Brushless Motor	15
Futaba R6004FF Receiver and Castle Creation Thunderbird 9 Electronic Speed Controller	13
2× Tahmazo TS-1002 Servo Motor	8
2× Driving Arm and Linkage	8
Tail Structure	11
Main Shaft	6
Main Gear	7
Middle Gear	5
Pinon Gear	1
Fuselage	15
Fasteners, Screws, Nuts and Wheel-Locks	7
Wings	35
Total Weight	160

### 3.2. Wing Construction

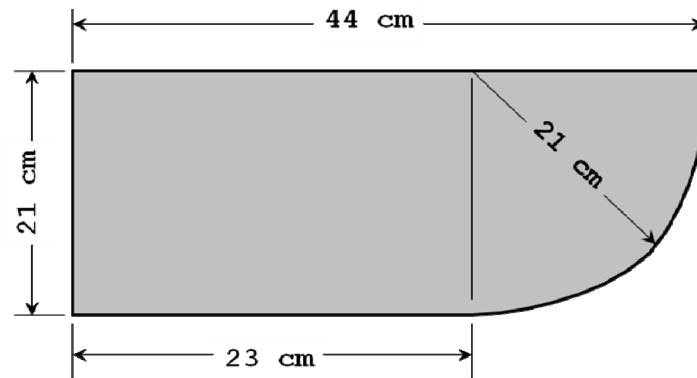
In the design and construction of any ornithopter, one of the most crucial components governing overall performance is the wings. Especially in a flapping-wing air vehicle, the wings are vital to flight performance aspects including endurance, speed, manoeuvrability, and many other useful behaviors. An effective ornithopter must have wings capable of generating both thrust, the force that propels the craft forward, and lift, the force, perpendicular to the direction of flight, that keeps the ornithopter airborne. These forces must be strong enough to counter the effects of drag and the weight of the ornithopter. The wings produce lift and thrust primarily due to their ability to change shape during the flapping motion. As the wing is accelerated, aerodynamic loading causes the wing to deform. The result is a fairly large camber change of the wing that results in the normally flat plate shape changing to an airfoil shape. The aerodynamic loading also produces large angles of attack that create thrust. When these two effects are combined, the airfoil wings are placed in a moving airstream, thus creating a flight sustaining lift force.

In order to achieve the desired flexibility and minimum weight, carbon fiber rods are used in the construction of the wings to provide a lightweight and stiff structure of spars, similar to the skeletal structure of a flying animal as shown in Figure 17. Ornithopters do not necessarily act like flying animals in flight. Typically flying animals have thin and cambered wings to produce lift and thrust. Ornithopters with thinner wings have a limited angle of attack but provide optimum minimum-drag performance for a single lift coefficient. The Pigeon Hawk has the closest characteristic as our ornithopter prototype, with the weight of 181.3 g and therefore the dimension of the constructed wing is as shown in Figure 18 [9].

There are two different materials used for the wings. The first material used is a polyester film made from stretched Polyethylene Terephthalate (PET). It has high tensile strength, and is commonly used in ornithopter flights. The second material used is Orcon, which is a better material than PET film. The Orcon is a strong and ultra-lightweight material. Unlike other plastic and mylar-like films, Orcon will not continue to tear even a small slit is detected.

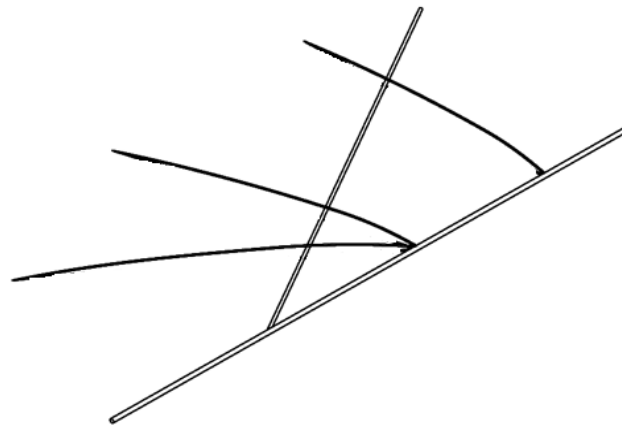


**Figure 17.** View of wing skeletal structure—one half.

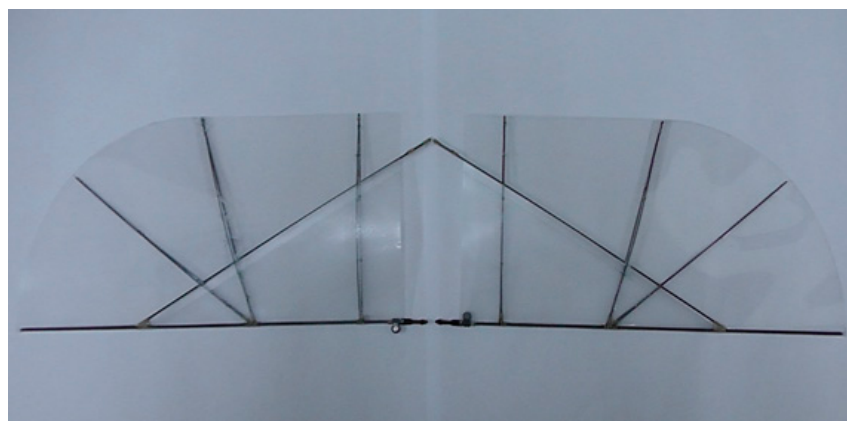


**Figure 18.** Dimension of the constructed wing [9].

Three different types of wings designs with various material and skeletal structure are fabricated. The first type is called PET cambered thin wing with a skeletal structure as shown in Figure 19. Figure 20 shows the PET cambered thin wing that is fabricated using the PET material.

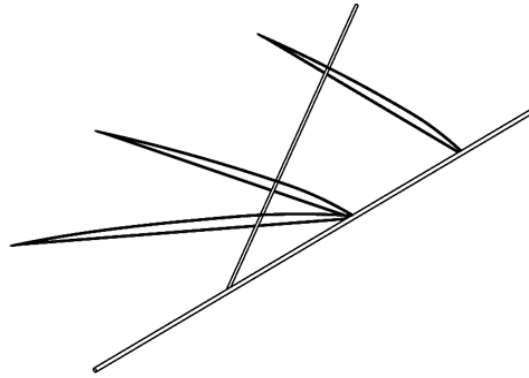


**Figure 19.** Skeletal structure for polyethylene terephthalate (PET) cambered thin wing (CAD Model).



**Figure 20.** PET cambered thin wing.

The second type of wing is called Orcon cambered thick wing with a skeletal structure as shown in Figure 21.



**Figure 21.** Skeletal structure for Orcon cambered thick wing (CAD model).

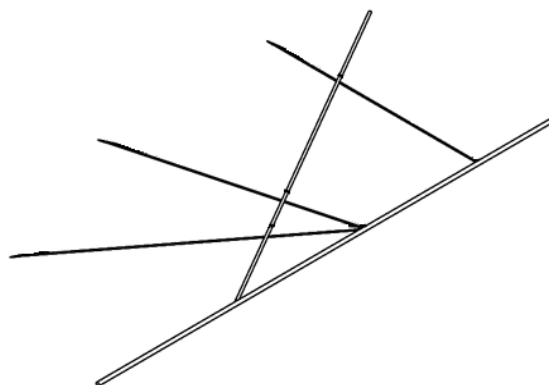
Figure 22 shows the Orcon cambered thick wing that is fabricated using the orcon material.



**Figure 22.** Orcon cambered thick wing.

The first two types of wings that are mentioned above are designed to have cambered due to the advantages of the increased lift-drag ratios and more desirable stall characteristics. It has a higher lift coefficient than the symmetrical airfoil. The top edge of the airfoil is shaped differently than the bottom edge, which changes the way air flows over it. This causes the air to move faster, which creates more lift.

The third type of wing is designed with a thin airfoil, called the Orcon flat wing with a skeletal structure as shown in Figure 23.



**Figure 23.** Skeletal structure for Orcon flat wing (CAD model).

The lift coefficient is sacrificed for a lighter weight. Lighter weight can produce smaller inertia of the wing and thus faster flapping speed, which will in turn increase air velocities. Figure 24 shows the Orcon flat wing which is fabricated using the orcon material.



**Figure 24.** Orcon Flat Wing.

### 3.3. Center of Gravity (CG)

The center of gravity may be defined as the average location of the mass distribution. It is a point in space where, for the purpose of various calculations, the entire mass of a body may be assumed to be concentrated. The center of gravity is an important point on an aircraft, which significantly affects the stability of the aircraft to fly safely. Therefore the center of gravity must fall within specified limits to ensure the ornithopter is stable. A simple method is adopted to determine the CG of the prototype ornithopter and that is to hang the prototype ornithopter from the ceiling using a kevlar thread as shown in Figure 25, the CG is obtained when the prototype ornithopter is equilibrium and balanced horizontally with the ground level.



**Figure 25.** Equilibrium position of the prototype ornithopter.



### 3.4. Lift and Thrust Force Measurement

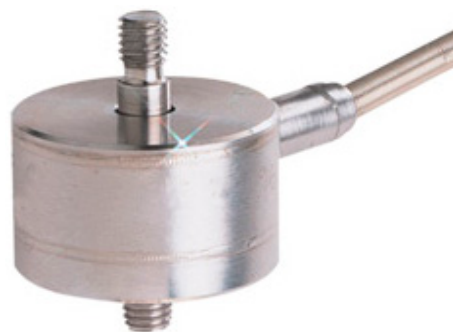
The experimental apparatus utilized by the authors include a force measuring load cell, a test assembly, and video recording is also taken of the mechanism with the various wing designs flapped. This section also presents the calibration of the load cell to ensure that the data measured is accurate. After the calibration is done, the data unit of the force measurement is converted from electrical signal into gram or Newton unit.

#### 3.4.1. Force Measuring Load Cell

The first and most important component of the force measurement system is the transducer. A load cell is a transducer that is used to convert the loads generated by the flapping wing into voltage signal for subsequent recording and processing. Through a mechanical arrangement, the force being sensed deforms a strain gauge. In most cases, four strain gages are used to obtain maximum sensitivity and temperature compensation. Two of the gauges are usually in tension, and two in compression. A strain gauge is a series of thin wire filaments wound in a serpentine fashion and placed in a Wheatstone Bridge configuration.

Voltage is supplied to the strain gauge and as a load is applied to the wire filaments, they will either elongate or shrink, changing the resistance in the wires. This variation in resistance results in different input and output voltages from the strain gauge. The difference in voltage is then used to calculate the strain.

It should be noted that the expected measured thrust and lift are very small, *i.e.*, around 0.1–1.0 N. The force measuring load cell utilized in this project is an Omega Engineering, LCMFD-20N load cell which is shown in Figure 26. This LCMFD-20N load cell is small in size and capable of providing highly accurate readings of a 2041 gram capacity. According to Omega Engineering's website ([www.omega.co.uk](http://www.omega.co.uk)) on technical specification, this LCMFD-20N load cell has its accuracy (linearity and hysteresis combined) and uncertainty of 0.15% FSO (Full-Scale Output), *i.e.*, 0.03 N (0.15% of 20 N). Since, the accuracy and uncertainty numbers are about 0.03 N (order of  $10^{-2}$ ), *i.e.*, about 1 order of magnitude less of the expected minimum measured thrust and lift values around 0.1–1.0 N (order of  $10^{-1}$  to 1), this is acceptable. In addition, this model is selected because of its high frequency resonant characteristics, minimal contamination from off-axis loads, and robust overloading tolerances.



**Figure 26.** Omega engineering, LCMFD-20N load cell.

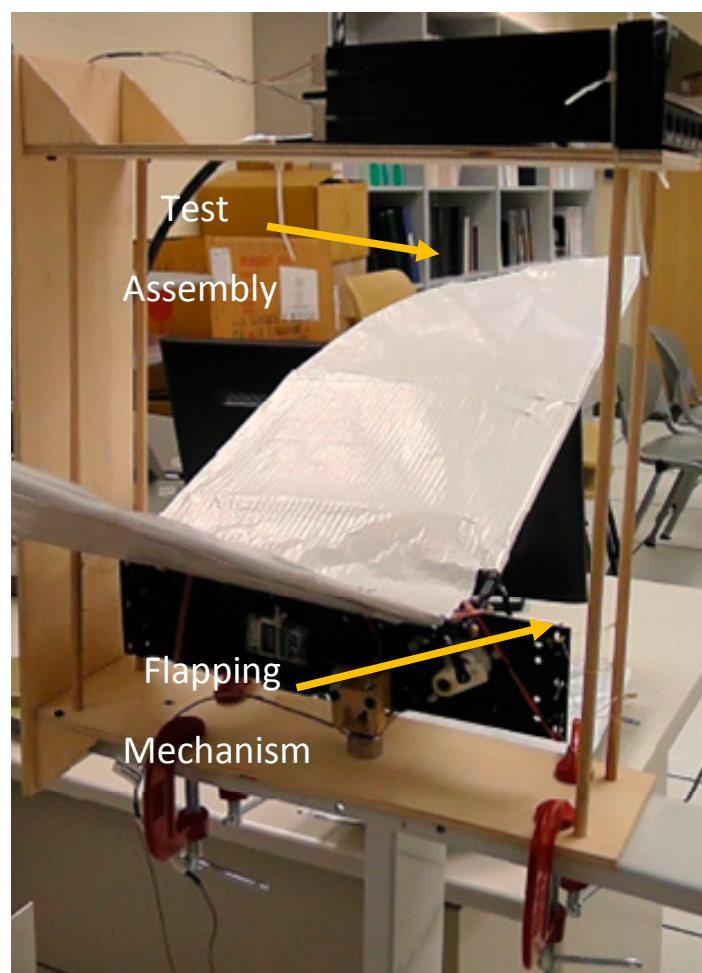
The load cell measures the forces when the prototype ornithopter flaps its wing on a force balance by converting these forces into electric signals, which will be displayed on the Omega DP41 digital panel.



### 3.4.2. Test Assembly

One of the challenges in this project is to design and build a test assembly made of pinewood in which the mechanism could flap and also measure both lift and thrust forces with the use of the load cell. A flapping mechanism in the test assembly can be seen in Figure 27. The position of the load cell is mounted to a fixed point on the test assembly, and by rotating the test assembly vertically together with the respective kevlar thread connected to the flapping mechanism, this testing assembly could measure the lift and thrust forces generated by the flapping mechanism. In addition, this only allow the flapping mechanism to move in either  $\pm X$  axis, which measures the thrust force, or  $\pm Y$  axis, which measures the lift force.

G-clamps are used to clamp the test assembly to a metal stand. This allows the wing tests to be done at a distance above ground so that we do not take into account the ground effects when the wings are flapping.



**Figure 27.** Test assembly and flapping mechanism.

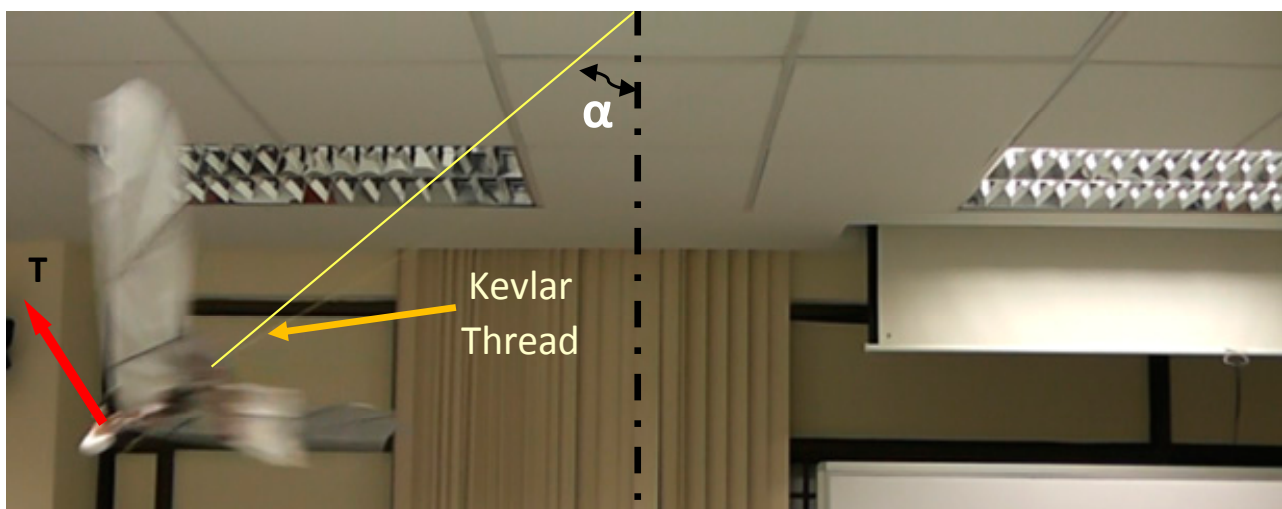
### 3.4.3. Lift and Thrust Forces Measurement Methods

As the objective of this project is to characterize the different lift and thrust generation performance from three different wing designs, namely PET cambered thin wing, Orcon cambered thick wing and Orcon flat wing, different thrust measurement techniques will be investigated. This section presents

the various methods for testing the lift and thrust generation of the prototype ornithopter. Kevlar thread is used extensively in this project as this innovative thread is light and its strength to weight ratio makes it five times as strong as steel. Kevlar thread does not break instantly but progressively, providing a non-catastrophic failure mode allowing a margin of safety.

#### Thrust Measurement by Swing Method

In this method, the prototype ornithopter is tightly attached to one end of the connecting kevlar thread and the other end is attached to the ceiling, making the body axis of the prototype ornithopter perpendicular to the connecting kevlar thread, as shown in Figure 28. The three different pairs of wings are tested using this method to measure the thrust produced.



**Figure 28.** Thrust measurement of swing method.

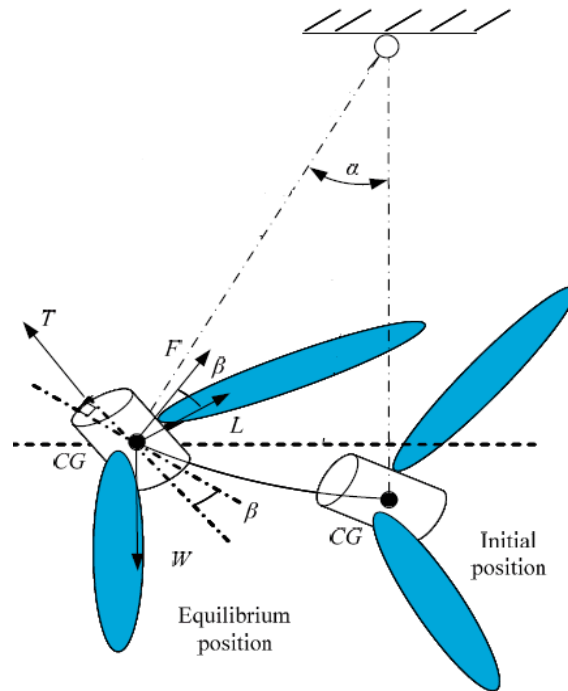
When the prototype ornithopter is excited, it produced thrust and moved to an equilibrium position. Assuming that all forces act at the center of gravity of the prototype ornithopter as shown in Figure 1, the equilibrium equation of the ornithopter after applying voltage is

$$T + F + L + W = 0 \quad (1)$$

where  $T$ ,  $F$ ,  $L$ ,  $W$  are the thrust, tension of the connecting kevlar thread, side force, and weight of the prototype ornithopter, respectively.  $T$  and  $L$  are two components of the resultant aerodynamic force projected on the ornithopter's body axis and the direction perpendicular to the ornithopter's axis, respectively. The equilibrium condition in vertical and horizontal directions can be modified as follows:

$$\begin{aligned} \sin(\alpha + \beta) + F\cos\alpha + L\cos(\alpha + \beta) - W &= 0 \\ T\cos(\alpha + \beta) - F\sin\alpha - L\sin(\alpha + \beta) &= 0 \end{aligned} \quad (2)$$

where  $\alpha$  and  $\beta$  are the swing angle of the connecting kevlar thread and the body angle of the ornithopter, respectively, as shown in the Figure 29.



**Figure 29.** Force diagram of swing test.

The body angle  $\beta$  is the angle between the ornithopter's body axis and the line perpendicular to the connecting kevlar thread. By eliminating the tension  $F$  in Equation (2), the thrust is calculated as follows:

$$T \cos \beta - L \sin \beta = W \sin \alpha \quad (3)$$

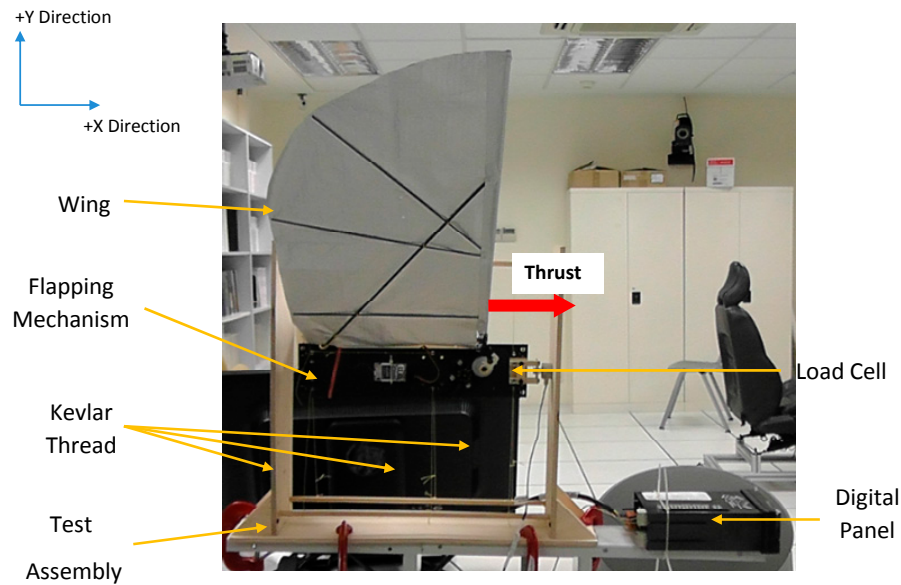
Hence, if the ornithopter's body axis is zero ( $\beta = 0^\circ$ ; *i.e.*, the body axis of the ornithopter is perpendicular to the connecting kevlar thread), the thrust at equilibrium can be calculated by:

$$T = W \sin \alpha \quad (4)$$

#### Load Cell Thrust Measurement Experiment Setup

In this thrust measurement setup, the mounting orientation of the load cell is shown in Figure 30. The load cell is fixed onto the test assembly in a stationary position. The front portion of the flapping mechanism is attached to the load cell while the top and bottom segments of the flapping mechanism are tied vertically to the test assembly with the use of kevlar thread. Thus, after mounting the pair of wings onto the flapping mechanism and turn on the motor, the flapping mechanism is constrained by the kevlar thread so that it is only capable of motion in the  $X$ -direction.

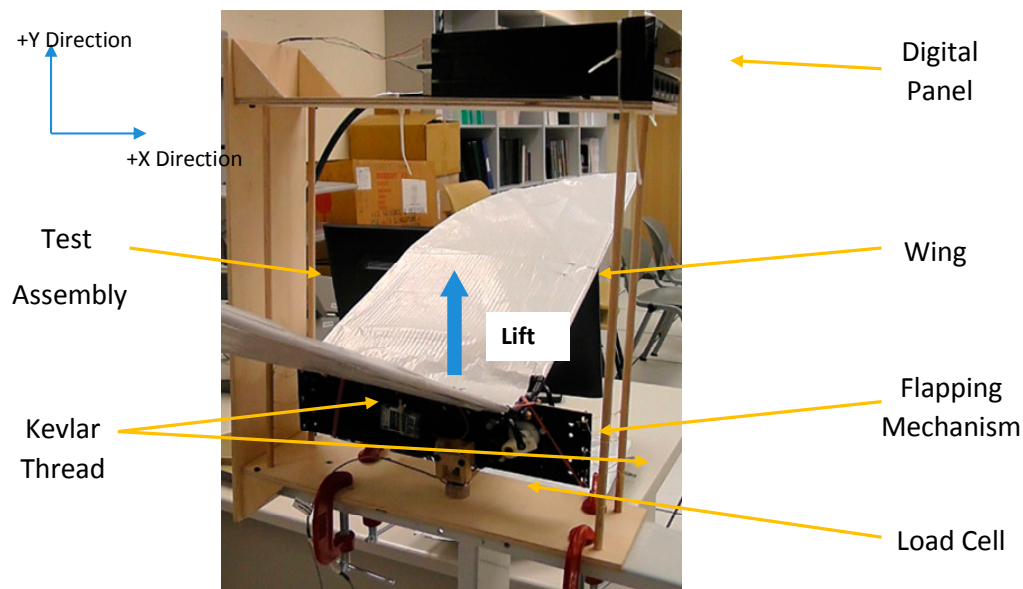
Motion in the  $+X$  direction represented thrust, while motion in the  $-X$  direction represented drag. As the flapping mechanism is moving in the  $+X$  direction, the load cell is under compression and therefore the readings displayed on the digital panel are negative values as discussed in Section 2 Load Cell Calibration. In this configuration, the thrust produced for the three different wing designs at three different flapping frequencies will be measured and then recorded in an Excel spreadsheet.



**Figure 30.** Side view of the load cell thrust measurement setup.

#### Load Cell Lift Measurement Experiment Setup

In the lift measurement setup, the test assembly is vertically rotated where the bottom part of the flapping mechanism is attached to the load cell as shown in Figure 31. The kevlar thread is tied horizontally at the front and back segments of the flapping mechanism, so that it is only capable of motion in the  $Y$  direction which the lift force is measured.



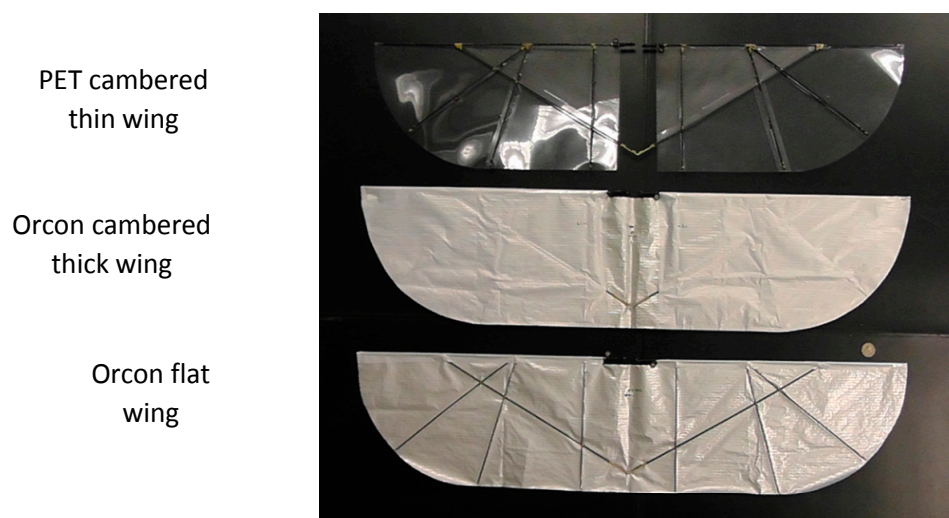
**Figure 31.** Load cell lift measurement setup.

As the flapping mechanism is moving in the  $Y$  direction, the load cell is under both compression and tension, which resulting in negative and positive values respectively displayed on the digital panel. This is due to the matter of fact that a flapping wing is an aerodynamic machine with two strokes, the upstroke and the downstroke. In this experiment, the lift produced for the three different wing

designs at three different flapping frequency will be measured and recorded in an Excel spreadsheet. Before recording the lift values during the flapping period, the weight of the flapping mechanism with the wings attached is measured at static condition. By subtracting the weight measured at static condition from all the values obtained during the flapping phase, the true lift force produced by the prototype is then recorded for analysis.

#### 3.4.4. Determination of Flapping Frequency

The flapping frequency of the wings when different amount of power is supplied to the motor is to be determined so as to make a comparison between the three different wing designs, as shown below (Figure 32), at the various flapping frequency.



**Figure 32.** Three different wing designs (bottom view).

The measurement of the flapping frequency of the prototype is achieved through the use of a digital camera. The digital camera is placed in front of the test assembly which allows the author to analyze how the wings are flapped in up- and down-stroke manner. The videos are then loaded into the Ulead software to process the captured video into frames. There are 25 frames for each second of the video recorded. Hence, by calculating the number of frames taken for the prototype to produce a full cycle consisting of upstroke and downstroke as shown in Figure 33, the period ( $T$ ) can be obtained using the following equation:

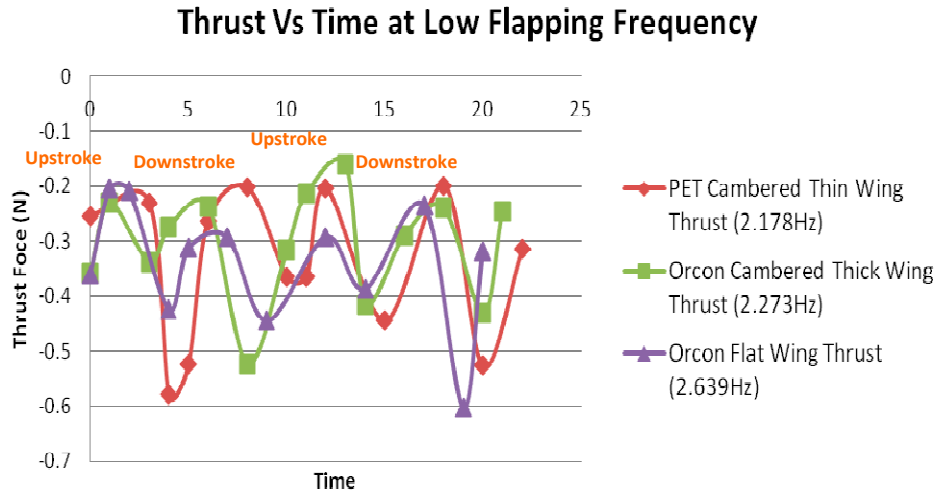
$$T = (1/25) \times \text{number of frames per full cycle} \quad (5)$$

Once the period is obtained, the flapping frequency ( $f$ ) is calculated as follows:

$$f = (1/T) \quad (6)$$

Hence, the flapping frequency can then be determined. The figure below shows the motion of the flapping wings in one full cycle.





**Figure 33.** Thrust against low flapping frequency (2 full cycles).

### 3.5. Lift and Thrust Force Results and Analysis

This section presents the findings of various experiments that the authors have conducted. The main element of this section is the results and analysis of evaluation of different wing designs. The average thrust and lift output are measured and compared to different wing designs to understand what styles of wings are more effective. The time-varying force profile can be examined to gain understanding into why particular wing designs prove more effective than others. The results of thrust and lift testing will be analyzed to establish any trends that explain performance differences and this section will be split into five different parts:

- Comparison of thrust generation performance between various flapping frequencies at a particular wing design.
- Comparison of thrust generation performance between wings of various designs against various flapping frequencies.
- Comparison of thrust performance between the load cell measuring setup and swing method.
- Comparison of lift generation performance between wings of various designs at different flapping frequencies.
- Investigation of lift generation between wings of different designs at various phase angles.

Thrust generation tests are done using the load cell setup method on the three different wing designs. For each wing design, three different flapping frequency starting from low, then medium and lastly to high are evaluated.

It is observed from Figure 33 that the thrust tends to produce four peaks. This is due to the motion of flapping wings. For each test, thrust performance is recorded in terms of two flapping cycles that consist of upstroke, downstroke, upstroke and downstroke motion, which explains the four peaks that are captured in the graph.

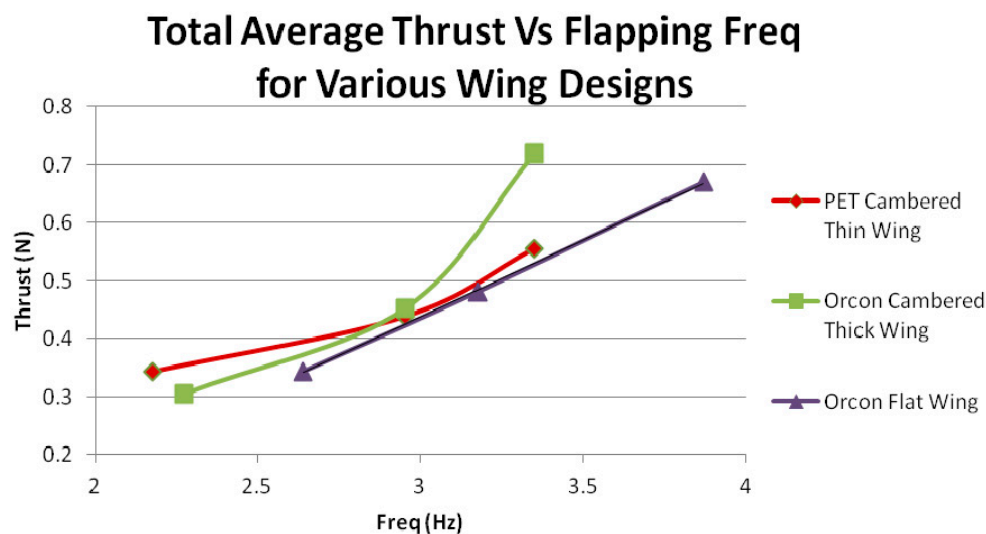
As mentioned in earlier, negative values are obtained when the load cell is under compression. Since the load cell is experiencing compression throughout the flapping period, it means that thrust is always being generated throughout the flapping cycle. More thrust is generated on the downstroke as compared to upstroke.



Apart from that, the slight periodic nature of the thrust is due to the inertial effects of the flapping mechanism. In the graphs above, the inertial effects have not been removed. When flapping with the test set up as described earlier, some inertial effects were present in the thrust direction. This is because the clearances between the flapping mechanism and the test assembly did not purely constrain the flapping in the vertical direction.

### 3.6. Thrust Comparison between Wings of Various Design

It can be observed from the results as shown in Figure 34 below that Orcon Flat wing has a better thrust generation performance as compared to the rest of the wing designs. It is interesting to note that the thrust for the Orcon flat wing resulted in a linear relationship when compared against the flapping frequency.



**Figure 34.** Total average thrust against flapping frequency for various wings.

The flapping wing mechanism constantly appears to produce thrust. The reason for this is that the wing is constantly changing its shape and angle of attack dynamically. On the downstroke the wing is pitched down, forcing air in the  $-Y$  direction, while in the upstroke the wing sweeps forward and up, minimizing the movement of air in the direction causing negative thrust. For both cases the thrust is relatively constant throughout the cycle. The magnitude of the thrust appears to slightly increase for the higher frequency.

Orcon Flat wing is much flexible than the other wing designs. Flexible wings can attain efficiency as more elastic the wing is, the more thrust produced and Orcon material is used to achieve a minimum weight. It is observed that PET cambered thin wing has a slightly slower flapping rate. It is because the weight of the flapping mechanism with that pair of wings on is 1.6 N, which weights the highest and tends to have more inertia. Hence, the motor will require more torque to drive the wings. With the same torque, heavier wing will have slightly slower flapping rate which results in a lesser thrust as compared to the Orcon Flat wing.

### 3.7. Thrust Comparison between Load Cell Setup and Swing Method

Swing angle  $\alpha$  is the angle between the initial position and equilibrium position of the prototype ornithopter, and is determined from digital camera images. As discussed, the thrust produced by the ornithopter can be calculated using Equation (5). Table 2 below shows the comparison of thrust results obtained from the various wing designs at three different flapping frequency.

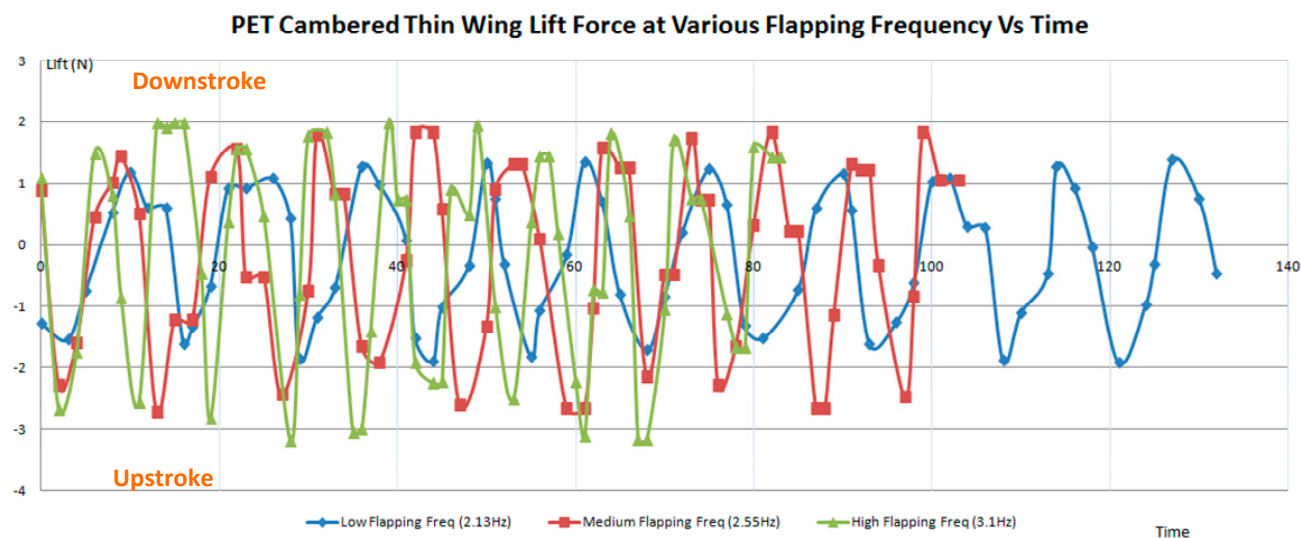
**Table 2.** Comparison of thrust between load cell and swing method.

Parameter	Flapping Mechanism with		
	PET Cambered Thin Wing	Orcon Cambered Thick Wing	Orcon Flat Wing
<b>At Low Flapping Frequency</b>			
Swing Angle ( $\alpha$ )	14°	12°	15°
Weight (N)	1.6	1.53	1.47
Swing Method Thrust $T = W \sin \alpha$	0.387	0.318	0.38
Load Cell Measurement Thrust (N)	0.343	0.305	0.344
<b>At Medium Flapping Frequency</b>			
Swing Angle ( $\alpha$ )	16°	18°	20°
Weight (N)	1.6	1.53	1.47
Swing Method Thrust $T = W \sin \alpha$	0.441	0.472	0.503
Load Cell Measurement Thrust (N)	0.438	0.452	0.481
<b>At High Flapping Frequency</b>			
Swing Angle ( $\alpha$ )	22°	28°	28°
Weight (N)	1.6	1.53	1.47
Swing Method Thrust $T = W \sin \alpha$	0.599	0.718	0.69
Load Cell Measurement Thrust (N)	0.555	0.719	0.67

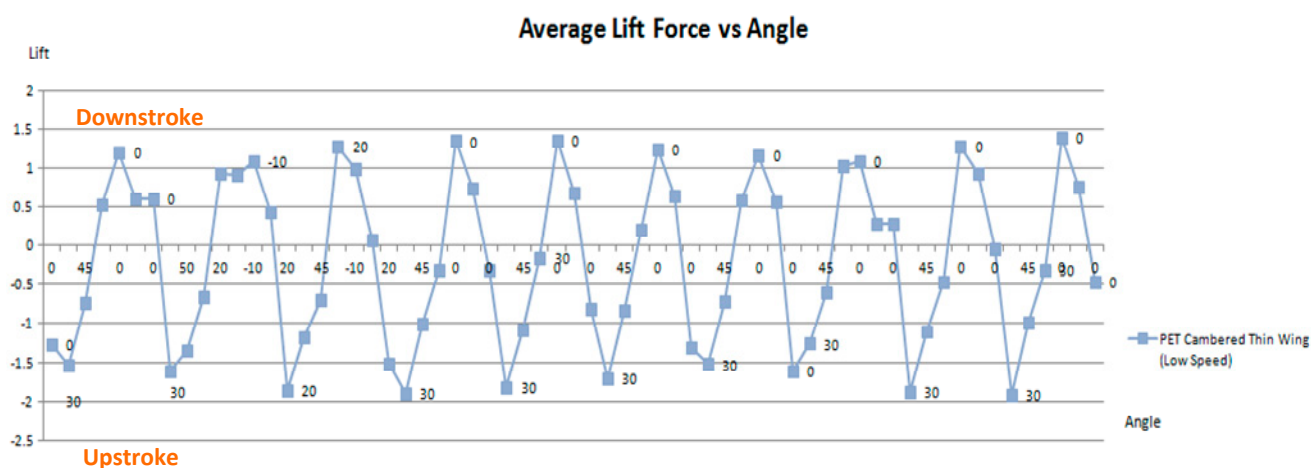
Lift generation tests are done using the load cell setup method as mentioned earlier on three different wing designs. Three sets of wings are tested at three different flapping frequency starting from low, then medium and lastly to high, and the lift force is compared at each speed. As the flapping mechanism is moving in the  $\pm Y$  direction, the load cell is under both compression and tension, which resulting in negative and positive values respectively displayed on the digital panel. This is due to the matter of fact that a flapping wing is an aerodynamic machine with two strokes, the upstroke and the downstroke. During the upstroke, the flapping mechanism will be “pushing” at the load cell, which results in a compression at the load cell and negative lift values will be displayed. During the downstroke, lift is generated and the flapping mechanism will be “pulling” the load cell, which results a tension at the load cell and positive lift values will be shown.

The first wing design to test is the PET cambered thin wing and the plots in Figure 20 shows the lift against a period of time of 10 full flapping cycle for three different flapping frequency.

It is observed from Figure 35 that as the flapping frequency goes higher, the lift is slightly increased. The average lift force generated during the low, medium and high flapping cycle are  $-0.196$ ,  $-0.14$  and  $-0.218$  N respectively. Figure 36 shows the amount of lift force generated at different angles for low flapping frequency.



**Figure 35.** PET cambered thin wing lift against time (10 full cycles).

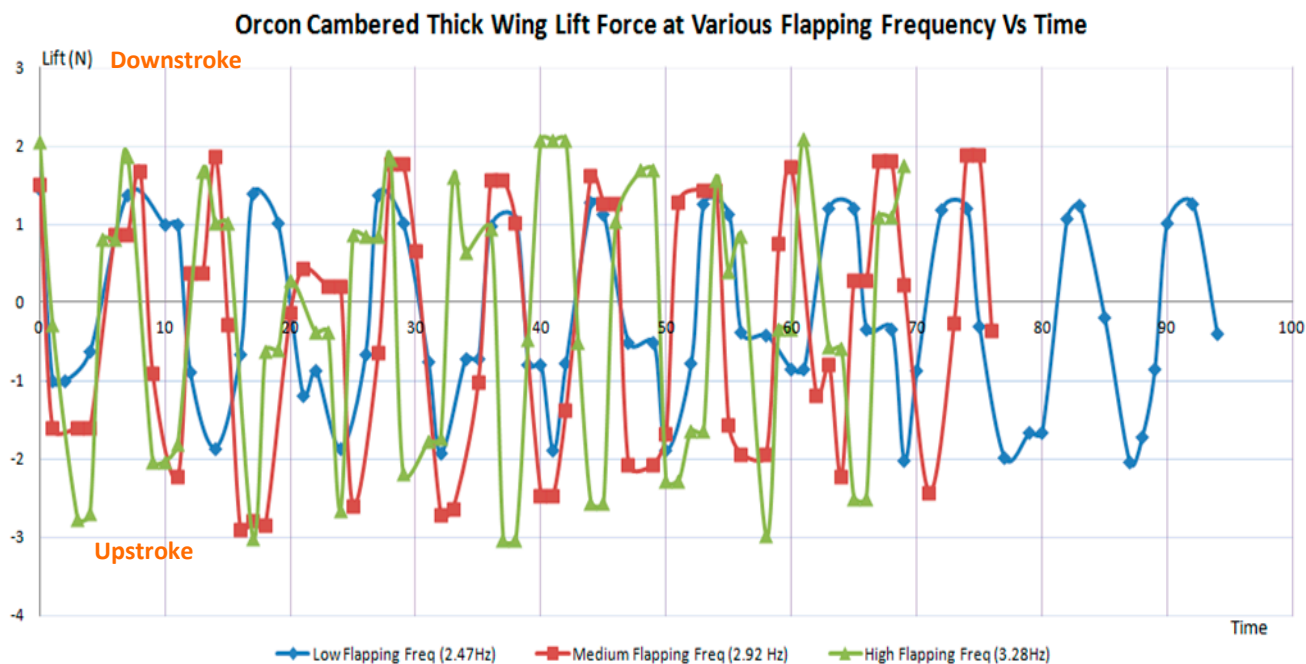


**Figure 36.** Average lift Force vs. angle (PET cambered thin wing).

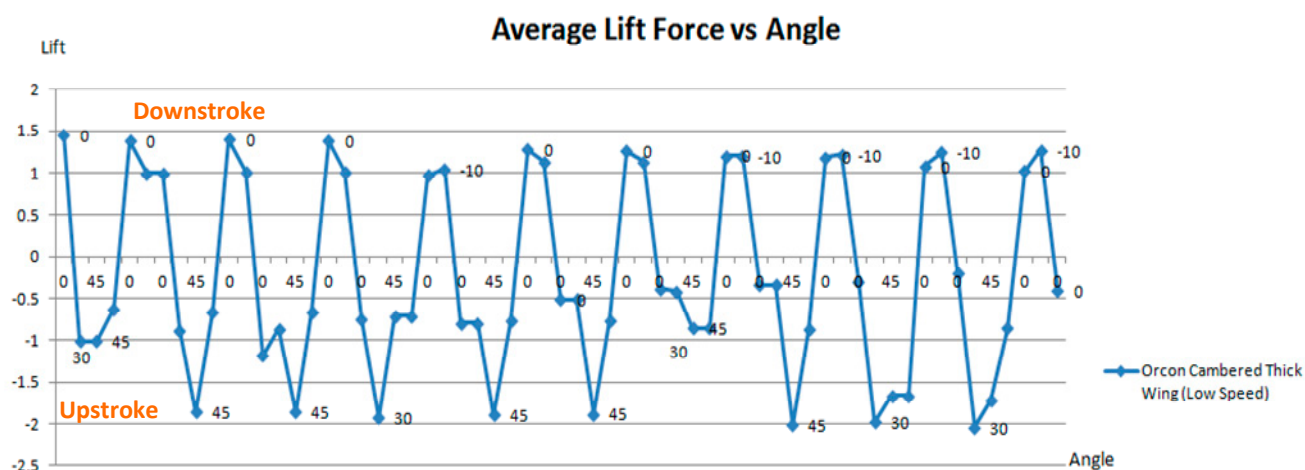
From the figure above, the results are consistent with the research made as discussed in earlier section that lift is generated at the down stroke, which reaches the maximum lift of 1.38 N at about  $0^\circ$  to  $-10^\circ$  angle. However, the minimum lift of  $-1.92$  N is reached when the flapping wing is on its upstroke. Note that the peak negative lift is greater than the peak positive lift, thus indicating that no net lift is generated.

The second wing design to test is the Orcon cambered thick wing and the plots in Figure 37 shows the lift against a period of time of 10 full flapping cycle for three different flapping frequency.

It is observed from the figure above that the faster the flapping frequency, the higher the lift is generated. The average lift force generated during the low, medium and high flapping cycle are  $-0.23$ ,  $-0.26$  and  $-0.3$  N, respectively. Therefore, it is also observed that faster the flapping frequency will result a higher negative lift force value. Figure 38 shows the amount of lift force generated at different angles for low flapping frequency.



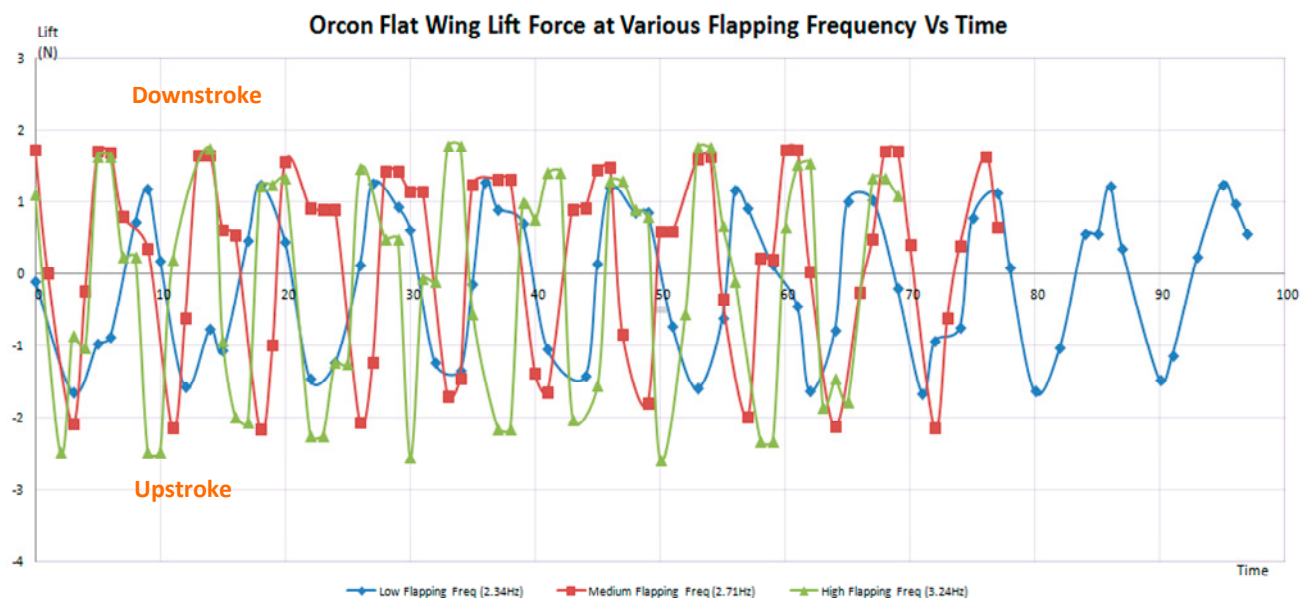
**Figure 37.** Orcon cambered thick wing lift against time (10 full cycles).



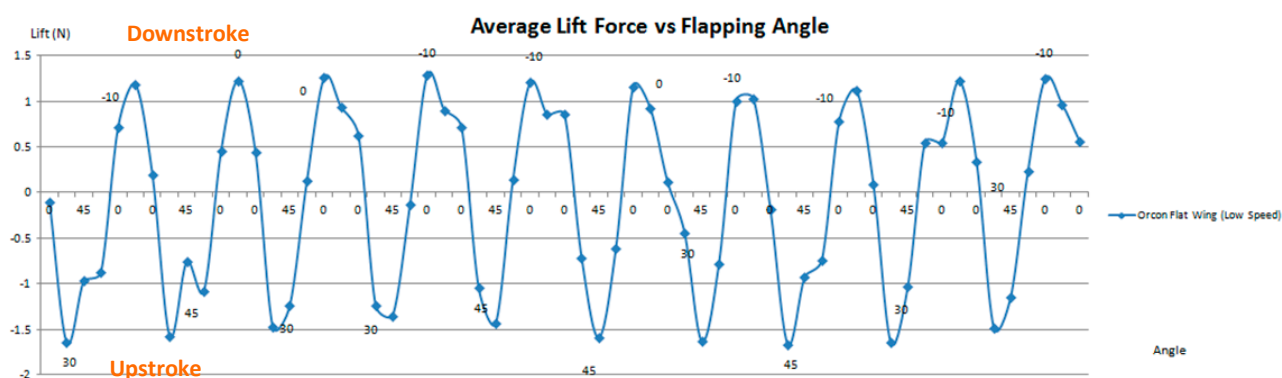
**Figure 38.** Average lift force vs. angle (Orcon cambered thick wing).

From the figure above, the result is similar to the PET cambered thin wing that lift is generated at the down stroke, which reaches the maximum lift of 1.46 N at about  $0^\circ$  to  $-10^\circ$  angle. However, the minimum lift of  $-2.04$  N is reached when the flapping wing is on its upstroke. Note that the peak negative lift is greater than the peak positive lift, thus indicating that no net lift is generated.

The third wing design to test is the Orcon flat wing and the plots in Figure 39 shows the lift against a period of time of 10 full flapping cycle for three different flapping frequency. From Figure 39, it is observed that faster the flapping frequency, the higher the lift is generated. The average lift force generated during the low, medium and high flapping cycle are  $-0.07$ ,  $0.26$  and  $-0.14$  N, respectively. It is interesting to find that a small amount of lift is generated at medium flapping frequency. Figure 40 shows the amount of lift force generated at different angles for low flapping frequency.



**Figure 39.** Orcon flat wing lift against time (10 full cycles).



**Figure 40.** Average lift force vs. angle (Orcon flat wing).

From these figures, the result is similar to the previous two tests that lift is generated at the downstroke, which reaches the maximum lift of 1.25 N at about  $0^\circ$  to  $-10^\circ$  angle. However, the minimum lift of  $-1.66$  N is reached when the flapping wing is on its upstroke. Note that the peak negative lift is greater than the peak positive lift, thus indicating that no net lift is generated.

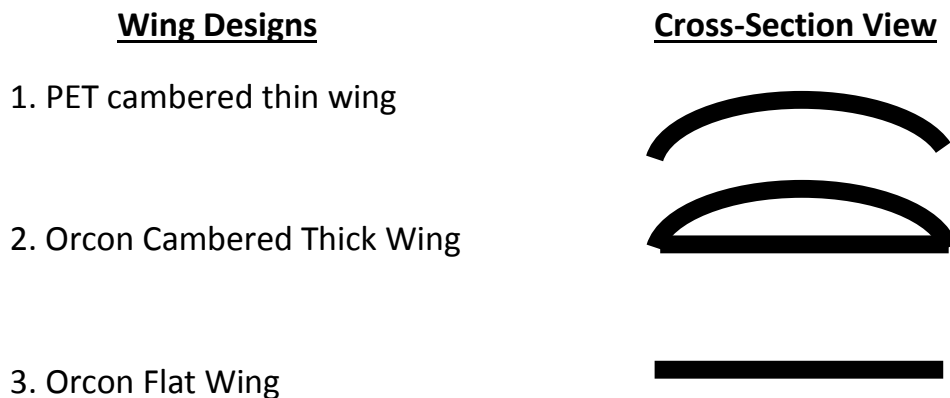
### 3.8. Comparison of Lift between Wings of Various Design

Table 3 shows that the only wing design that produced lift is the Orcon flat wing flapping at medium speed. However, most of the data obtained in this lift test has showed that lift forces produced are generally negative. This means that the prototype ornithopter will not be able to sustain a flight in the air. The Orcon flat wing has shown potential in generating lift with the most lift produced as compared to the rest of the wings. This may be due to the advantage of its light weight and flexibility of wing design. Having a lighter weight will actually allow the flapping mechanism to flap the wings faster. Therefore, improvements and modification are to be made to the wing design in order to produce the sufficient lift to sustain a flight.

**Table 3.** Comparison of lift forces.

Type	Low Speed (N)	Medium Speed (N)	High Speed (N)
PET Cambered Thin Wing	−0.196	−0.140	−0.218
Orcon Cambered Thick Wing	−0.225	−0.256	−0.305
Orcon Flat Wing	−0.075	0.264	−0.142

Figure 41 below shows the cross section view of the three various wing designs.

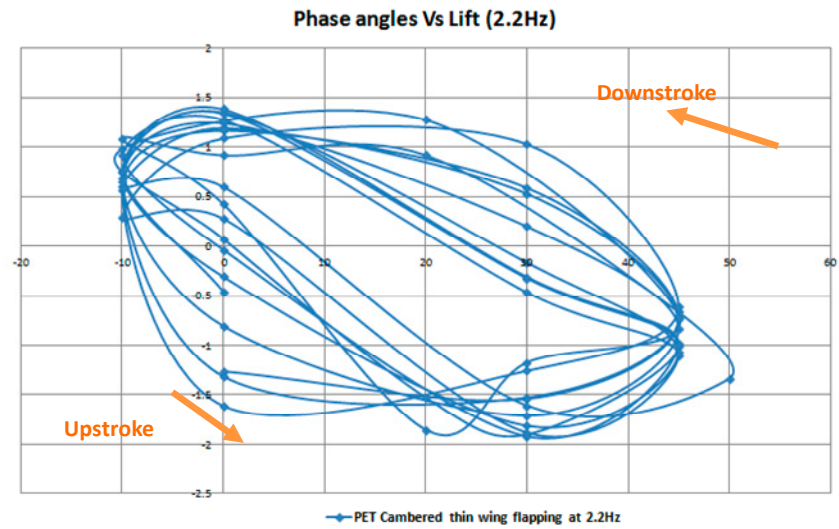
**Figure 41.** Cross-section view of the three different wing designs.

The next few figures (Figures 42–46) will show the investigation of phase angles against the lift generation for the three different wing designs at various flapping frequency. The first wing design to be discussed is the PET cambered thin wing, followed by the Orcon cambered thick wing and lastly the Orcon flat wing.

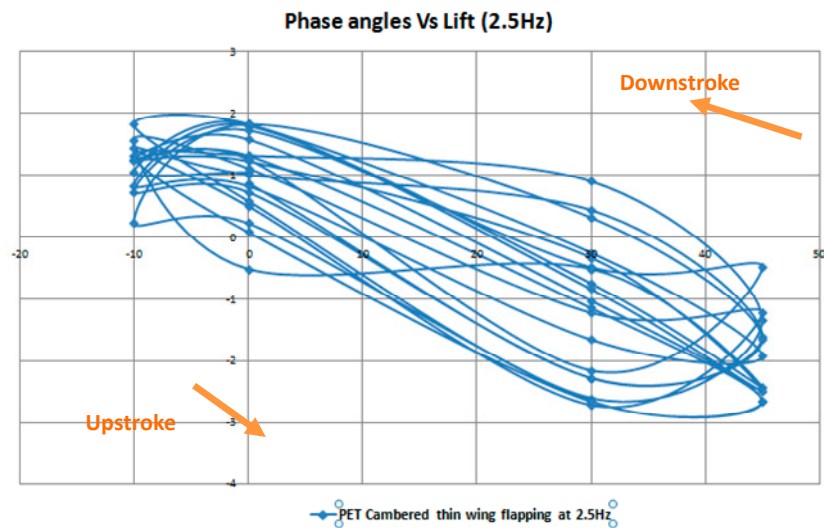
It is observed from the previous figures that as the flapping frequency increases, there is a transition from a “smooth circular” graph to a “figure-of-eight” graph. An example of the lift generation from the Orcon flat wing flapping at the medium frequency graph as shown below, during the initial downstroke, there is a sudden rise in lift until it reaches its peak. When the lift reaches its peak somewhere around  $30^\circ$ , it is noticed that constant lift is produced until it reaches at an angle of  $0^\circ$ . This could be the leading edge vortex (LEV) that causes it. It appears that LEV can enhance lift by attaching the bounded vortex core to the leading edge during wing translation. The vortex, formed roughly parallel to the leading edge of the wing, is trapped by the airflow and remains fixed to the upper surface of the wing. As air flows around the leading edge, it flows over the trapped vortex and is pulled in and down to generate the lift.

There are two routes that can be seen from the graphs, the first route which is the positive angle transit to negative angle (Downstroke) and the second route is the reverse of the first route (Upstroke). From Figure 46, the net lift can be easily seen by looking at the difference between the two routes.

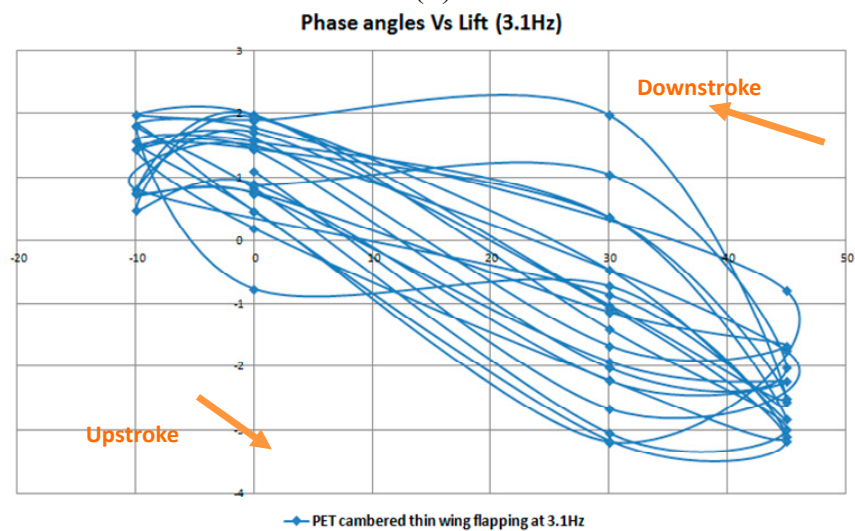




(a)

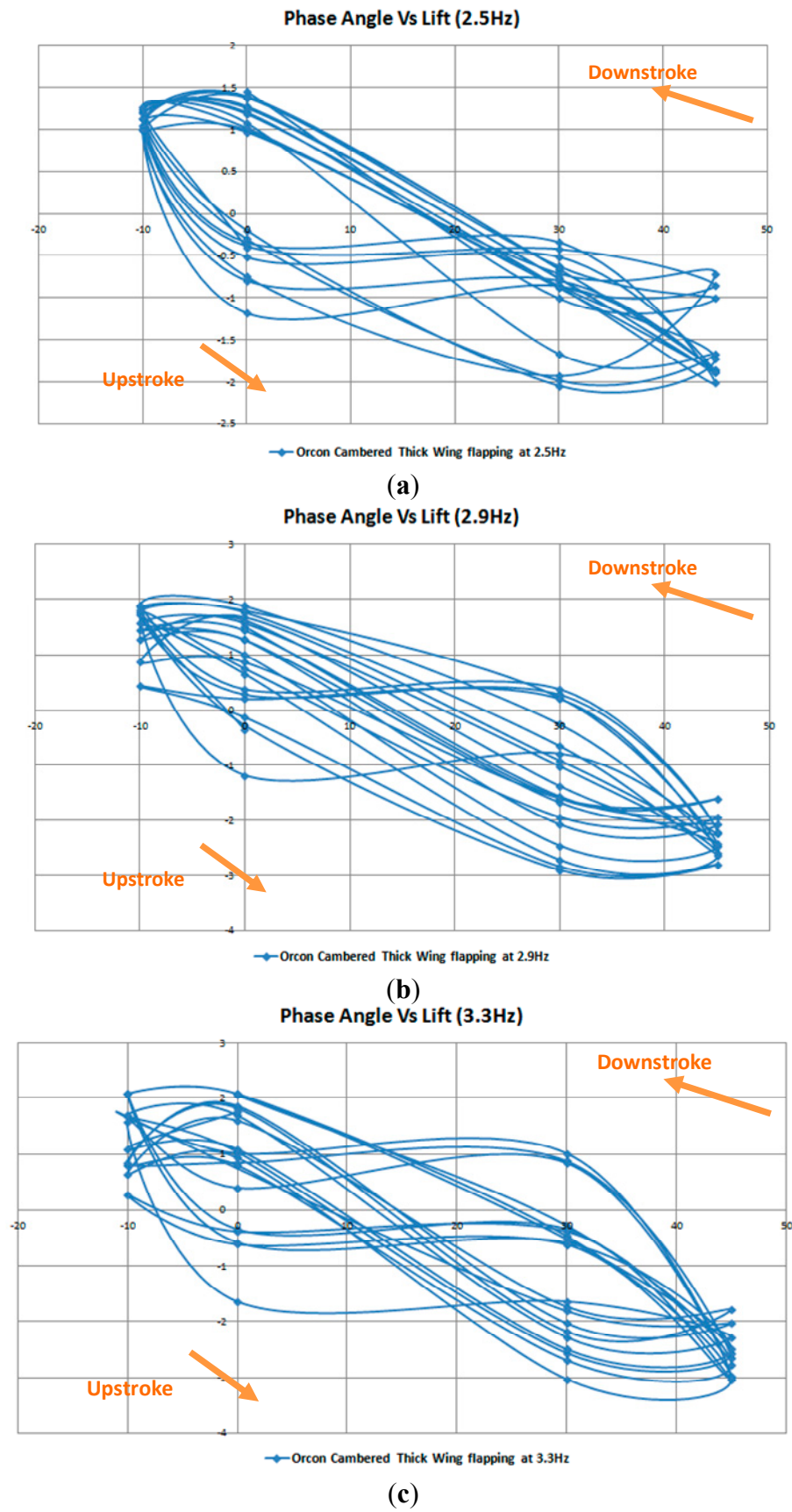


(b)

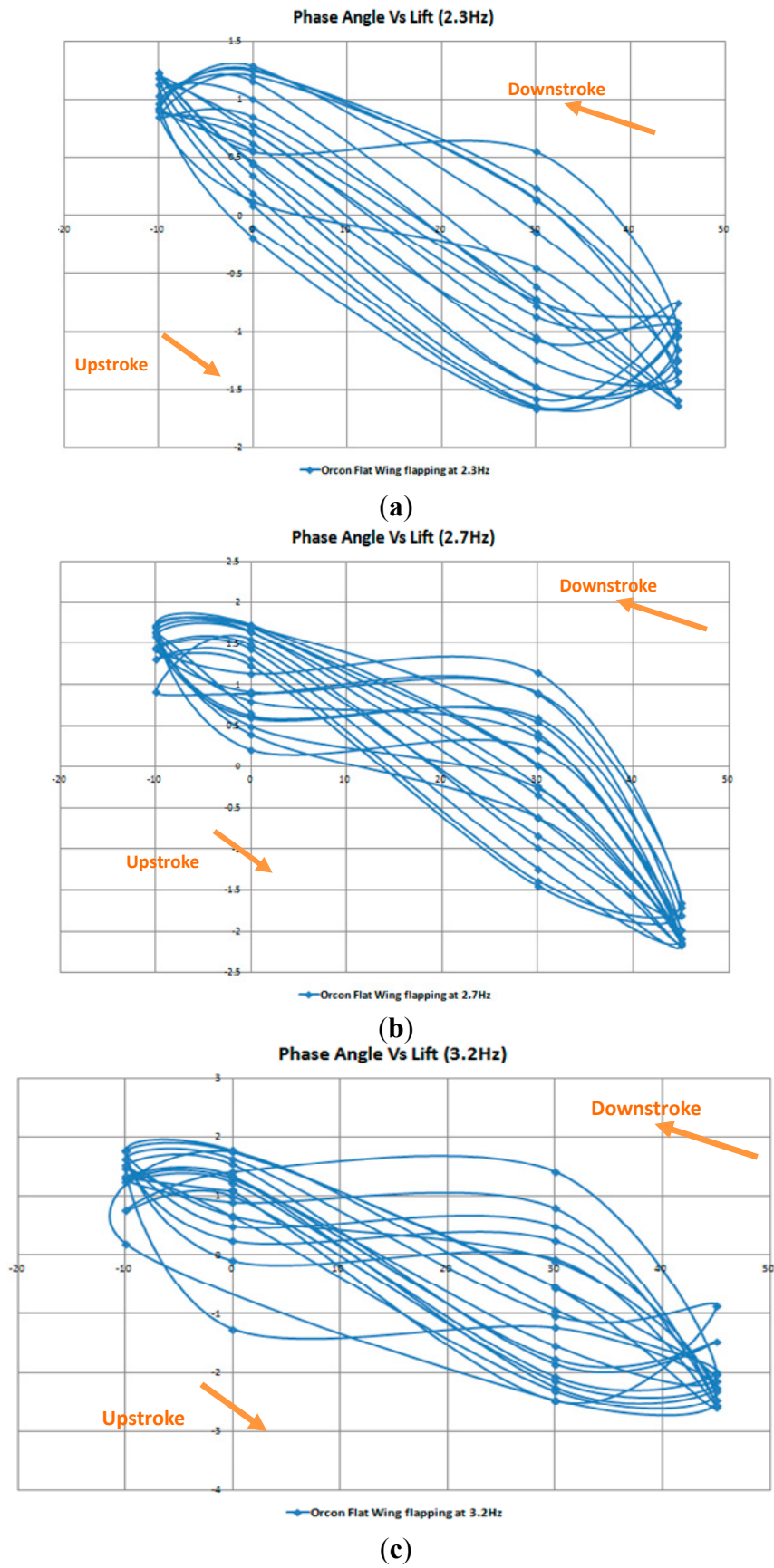


(c)

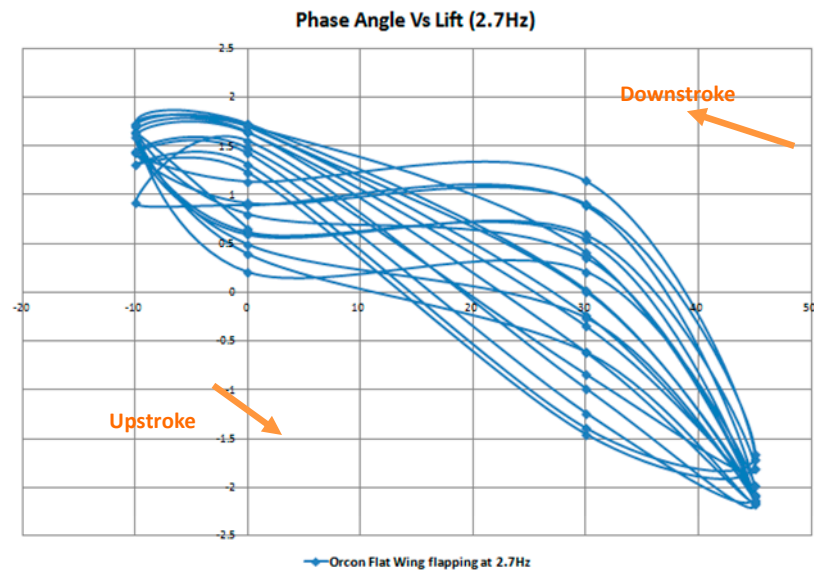
**Figure 42.** Phase angle vs. lift for PET cambered thin wing. (a) Low speed; (b) medium speed; (c) high speed.



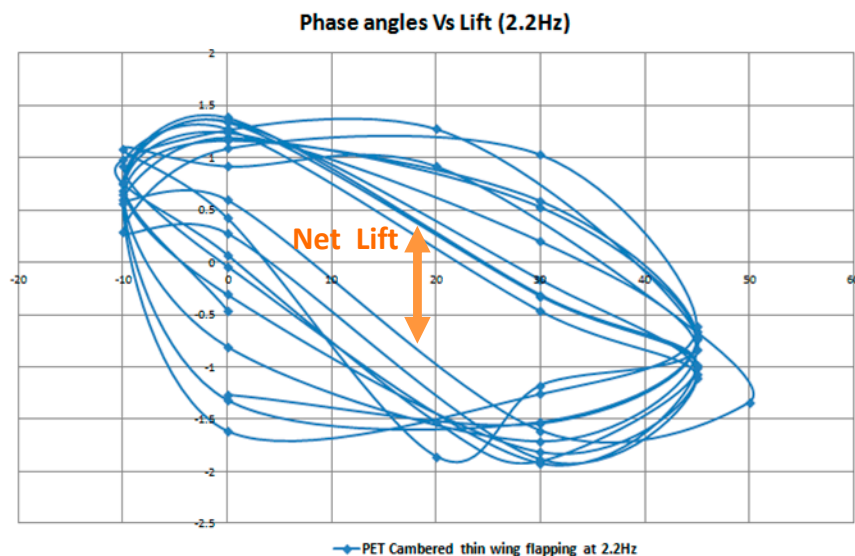
**Figure 43.** Phase angle vs. lift for Orcon cambered thick wing. (a) Low speed; (b) medium speed; (c) high speed.



**Figure 44.** Phase angle vs. lift for Orcon flat wing. (a) Low speed; (b) medium speed; (c) high speed.



**Figure 45.** Phase angle vs. lift for Orcon flat wing flapping at 2.7 Hz.



**Figure 46.** Net lift generation.

#### 4. UGS Flapping Wing MAV Prototypes 2 and 3

With the lesson learnt on the materials from the first prototype obtained in the previous section, and from other literatures ([29–33]), we designed, built and flew another two flapping wing MAVs using fabrication method such as laser cutting and Rapid Prototyping.

##### 4.1. Prototype 2

Our flapping wing MAV would be based on an albatross-like design. In addition to the results shown above with inspiration from other ornithopter-like MAVs (e.g., Delfly [34,35]). We introduced additional design criteria, e.g., it has to be lightweight, simple and yet strong enough to withstand the stress of the flapping motion and the crash landings during test flight. Simplicity is the key here as most of the components that would be used would be from hobby shops.

#### 4.1.1. Flapping Wing Mechanism

The flapping wing mechanism function is to convert the motor's rotary motion into flapping motion. It is the most important component of the MAV thus much research was done to assess the many different designs available. Generally the mechanism design is about the same to each other with only slight modifications.

##### Staggered Crank Design

The staggered crank design in Figure 47 is the most basic of the flapping wing design [36]). The connector rods are staggered in a measured distance and angle to ensure that the left and right wing are flapping symmetrically. This design is favoured by a hobbyist who wants to attempt to make their own Ornithopter using household items. Modifications have to be made so that the motor can be used instead of a rubber band as its power source.



**Figure 47.** Staggered crank.

##### Single Gear Crank Design

The single gear crank design in Figure 48 taken from University of California Biomimetic Millisystems Lab [37], looks simple however it is more complicated than it seems. Figure 48 shows the wings at the same level. The center point where the connector rod and the wing hinges are connected to each other has to expand and contract as the mechanism flaps. Contracting and expanding at a very high frequency could result in component failure.

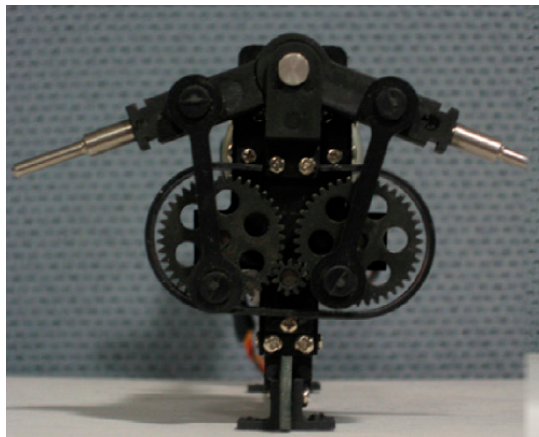


**Figure 48.** Single gear crank.



### Dual Gear Crank Design

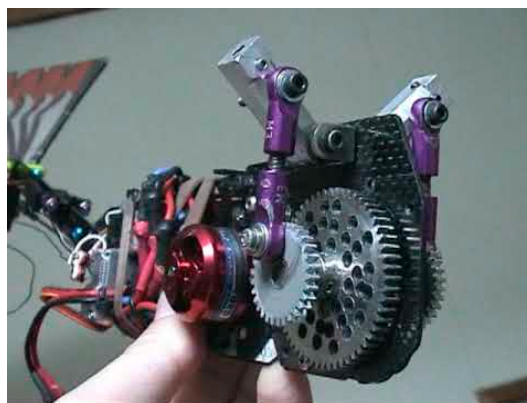
Figure 49 shows the dual gear crank design from similarly used in the Festo's SmartBird [38]. It features two gears that controls each wing hinges separately. There are different variation to the drivetrain design. The one shown in Figure 49, uses the pinion wheel to drive both the secondary gears. The secondary gears will rotate in the same direction with each other. In the other design, the pinion gear rotates the secondary gear and this secondary gear rotates another secondary gear. The secondary gears would rotate counter clockwise to each other. This design is much simpler to implement and reduce the wing symmetry misalignment.



**Figure 49.** Dual gear crank.

### Transverse Shaft

The transverse shaft design shown in Figure 50 is the other variation of flapping mechanism from [39] which allows for the most symmetrical flap, however, it is the heaviest and the most complicated design. The rotating gears and the flapping wings are not in the same plane thus the connector rod has to be able to rotate. The connector rod has a ball bearing inside and this adds weight to just the component itself. The number of gears used in this design is more than any other design. The transverse shaft design is usually used for a bigger MAV design where weight could be overcome by large wings.



**Figure 50.** Transverse Shaft.

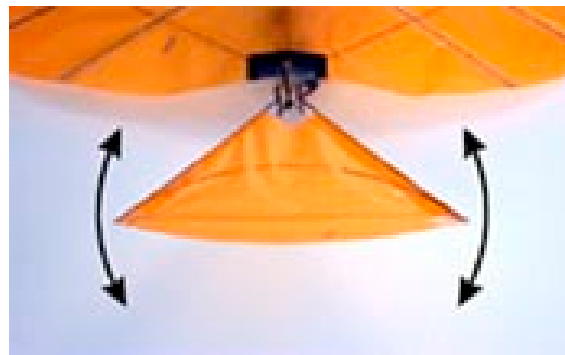


#### 4.1.2. Tail

The tail design varies with its intended use. Some of the design uses it only for stability but in most cases they are used for control as well. For stability, the tail is tilted upwards so that it the downward force of the tail would force the nose to pitch up. The angle is typically around  $15^\circ$  or less. For control the more common designs implemented are the swinging tail (Figure 51) and the tilting tail (Figure 52) due to their simplicity. The swinging tail works by causing a rolling moment to when it swings to either side. The tilting tail works like a rudder, when it tilts to the right it causes the MAV to yaw to the right. A horizontal stabilizer tail design unlike the other two designs could provide additional control. It can act as an elevon, providing pitch and roll control. However, this design requires two servos to be used and a more complicated design.



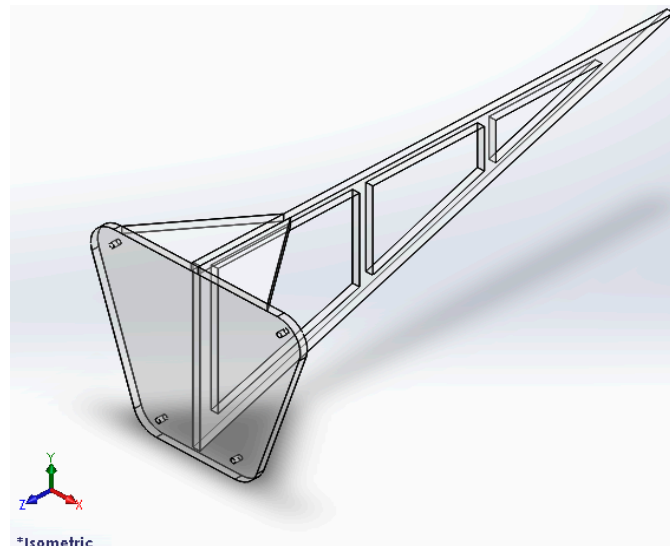
**Figure 51.** Swinging tail.



**Figure 52.** Tilting tail.

#### 4.1.3. Body

The body is the part where the components like the electronic speed controller, the receiver and the battery is located. The body also has to hold all the components from moving around too much. This is to prevent the shifting of the center of gravity of the MAV. The components would each be taped separately and then hooked to the body by Velcro tape. As the design would not require much space the body design could be hollowed. Figure 53 shows the body design with holes in them. This significantly reduces the total weight of the body. The body design had to be glued to the flapping mechanism at a  $90^\circ$  angle. Small triangles were added in between them as a support structure to prevent the body and the flapping wing mechanism from snapping off.



**Figure 53.** Body design.

#### 4.1.4. Gear and Motor Selection

The gear design was dependent on the motor that is going to be used. The motor rating affects the gear ratio which then affects the flapping frequency. The motor that is used is a brushless outrunner motor. Outrunner motors have lower KV ratings meaning they have more torque but less speed. More torque is needed than speed for this project as the motors have to turn the gears to flap. The motor also needed a front mount so that it could be mounted easily to the flapping mechanism frame instead of a separate mount just for the motor. This narrows down to two motors as shown on the Table 4.

**Table 4.** Motor specification comparison.

Specification/Motor	Motor 1	Motor 2
Motor Rating (KV)	1200	2800
Load Speed (rpm)	5800	8350
Voltage (V)	11.1	4
Weight (g)	38	25

Motor 2 was chosen as it was lighter and requires lesser voltage. Voltage is linked to the number of cells that the Lithium–Polymer (Li–Po) batteries has and the rating of Electronic Speed Controller (ESC). Each cell on a battery is 3.7 V so the higher the voltage the heavier the battery. It is the same for ESCs, higher ratings means bigger and heavier ESCs.

#### 4.1.5. Fabrication and Material

There were three materials being considered initially: Carbon fiber, balsa wood and acrylic. The first material of choice was to use carbon fiber due to it being strong and light. As it turns out, CO<sub>2</sub> laser cutting a carbon fiber sheet would burn the material. Balsa wood is very light and easy to cut; however, due to complex design of the MAV it was decided that it was not a suitable material. Hence, acrylic was selected. Acrylic is not as light and strong as carbon fiber however it can use CO<sub>2</sub> laser cutting machine to do precision cutting.

#### 4.1.6. CAD Design Dimensions

In order to find out the total dimensions and the weight that is allowed for flight, a lift equation was used. Certain assumptions made before using this equation are as follows:

1. The resulting lift would be higher in reality due to neglecting other flapping wing effects that contribute to lift when flapping.
2. The coefficient of lift is independent of the location on the wing and time.

From the assumptions made the equation for a rectangular wing shaped (of the same area) lift could be expanded to [38,40]:

$$L = \varphi_0^2 \cdot \pi^2 \cdot f^2 \cdot C_L \cdot \rho \cdot c_0 \cdot l^3 \cdot \frac{1}{3} \quad (7)$$

where  $\varphi_0$  is flapping angle,  $f$  is the flapping frequency,  $c_0$  is the chord length and  $l$  is the wing span length.  $C_L$  is obtained from the CFD results in the previous section (Section 2). This equation is to be used as a rough estimate so that the dimensions and weight of the MAV could be measured. Table 5 shows the results from using the equation.

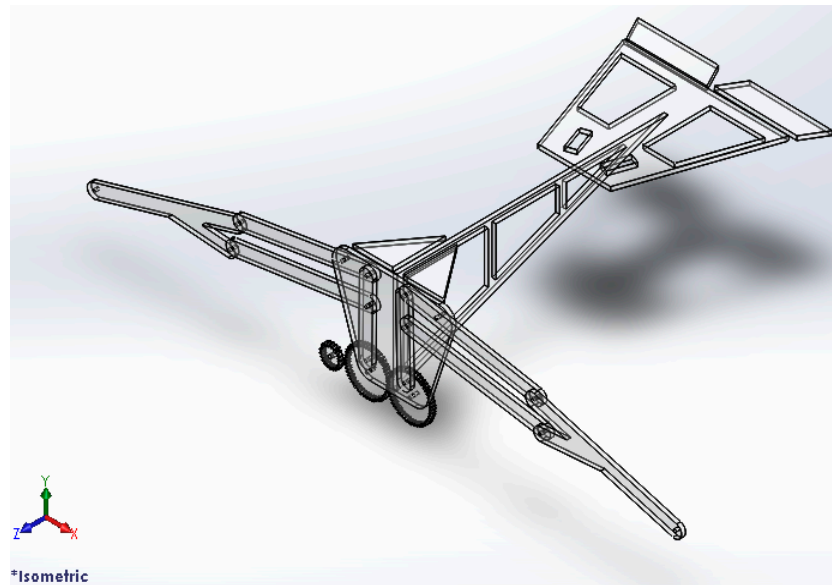
**Table 5.** Approximated lift generated.

Parameters	Values	Unit
Flapping Amplitude	70	deg
Flapping Frequency	6.5	Hz
Lift Coefficient	0.8	
Air Density	1.225	kg/m <sup>3</sup>
Chord Length	0.13	m
Wing Span	0.3	m
Lift	1.684	N

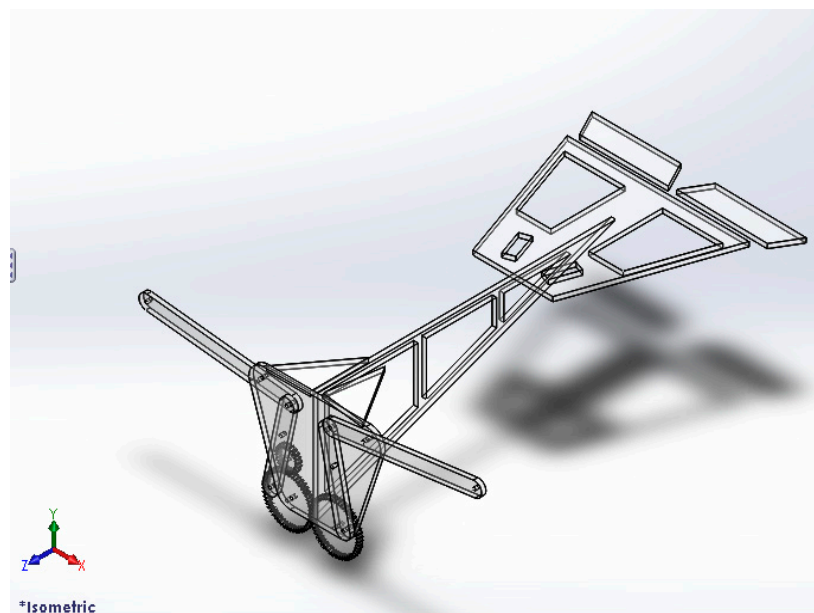
#### CAD Design

Using the dimensions above and the design criteria, a CAD design using SolidWorks was modelled. It would incorporate a dual gear crank and a horizontal stabilizer tail design. The dual gear crank was the simplest design with not much wing symmetry misalignment. The horizontal stabilizer tail design was chosen as it could provide both pitch and roll control. The initial design showed in Figure 54 featured an articulated wing. This design was not used as there were too many moving parts in the design and may complicate things. Therefore the chosen design is the one shown in Figure 55.

The total weight of the MAV was measured using one of the features in SolidWorks. Now the total weight of the MAV plus the components could be compared to the lift equation result. Table 6 shows the sum of all the component weights. The two measurements show that the weight of the MAV is below the total lift generated. An image of the assembled MAV is shown in Figure 56.



**Figure 54.** Articulated flapping wing.



**Figure 55.** Single flapping wing.

**Table 6.** Total weight of micro air vehicles (MAV).

Components	Values	Unit
Brushless Outrunner Motor	25	g
Radio Receiver	11.5	g
Servos	9	g
Li-Po Battery	15	g
Electronic Speed Controller	10	g
MAV Design	80.24	g
Total Weight	150.74	g



**Figure 56.** Assembled flapping wing MAV.

#### 4.2. Prototype 3

Learning from the prototype 2, some design considerations were made, *i.e.*, (1) the flapping mechanism needs to be more simplified; (2) the number of moving parts need to be reduced; (3) the overall design has to be much smaller to reduce weight; and (4) changing the tail design to either a tilting or swinging tail would reduce the number of servos used which would reduce weight.

##### 4.2.1. Flapping Wing Mechanism

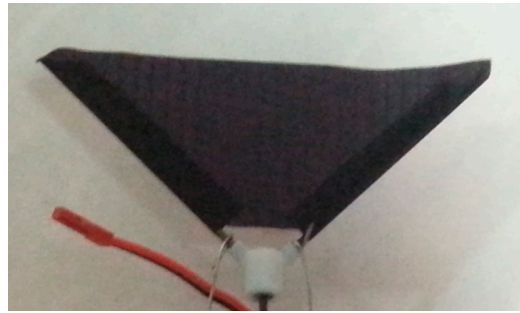
The flapping wing mechanism for prototype 2 had too many moving parts and was not simplified enough. A simpler design was needed and thus another look at the single gear crank was taken. The design idea was to shift its fixed pivot point from being at the center of the wing to it being at the end of the two wing joints. Figure 57 shows this design. The changes made to its pivot point made the flapping mechanism worked properly. A simulation test was done using the software and it showed that it could hold at high frequency flapping and the flapping movement is synchronized.



**Figure 57.** Prototype 2 flapping mechanism.

##### 4.2.2. Tail

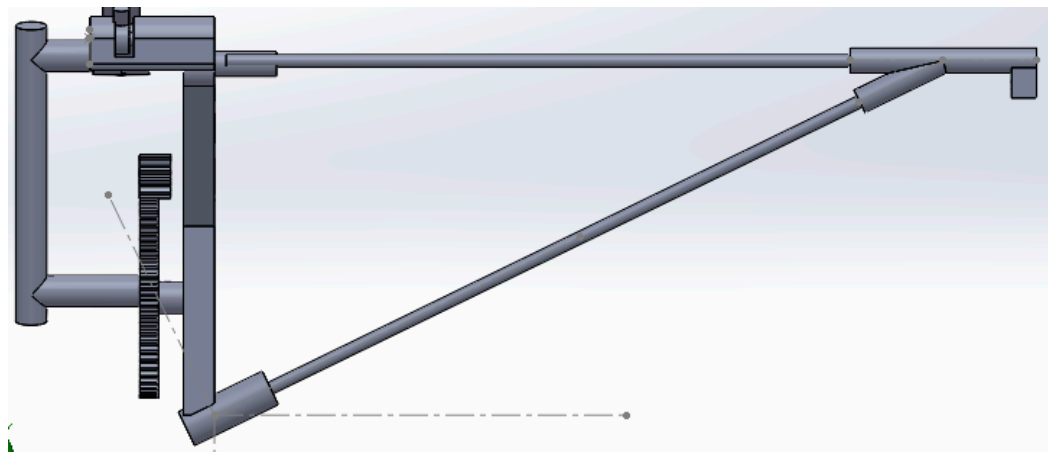
The previous prototype was using an elevon tail design which could provide pitch and roll control however it requires two servos to be used. For weight reduction and simplicity sake, a simple tilting tail would be used instead. The tail frame would be made up of carbon rods which would be fixed to the tail piece and covered with Ripstop. The tail piece has a ball bearing inside it so that the tail could tilt easily. Figure 58 shows the tilting tail design.



**Figure 58.** Prototype 2 tail.

#### 4.2.3. Body

In previous design the body was made out of acrylic and had to be solvent weld together. The design was simpler to implement however it was bearing a lot of weight. In order to reduce more weight, carbon rods would be connected to the front piece and the tail piece to form a rigid triangle frame. The frame would then be covered with Ripstop and Velcro tape to secure the components to the platform. Figure 59 shows the CAD design of the body.



**Figure 59.** Prototype 2 body.

#### 4.2.4. Gear and Motor Selection

The new flapping mechanism uses only two gears (Table 7). This allows more fine tuning to the gears which allowed a gear ration of 5.5:1. This was acceptable as the newer motor has a slower load speed but higher torque. The new gear is specially hollowed at the center for a ball bearing to be inserted so that it can spin freely around the connecting part of the front piece.

**Table 7.** Motor specification comparison.

Specification/Motor	Motor 1	Motor 2
Motor Rating (KV)	1700	2800
Load Speed (rpm)	7800	8350
Voltage (V)	7	4
Weight (g)	20	25



#### 4.2.5. Fabrication and Material

In the first design, a laser cutting machine was used. For this second prototype, a Rapid Prototyping Machine or also known as a 3D printer would be used. A 3D printer allows for more freedom of design. An extruded part could be combined during the design process easily, compared to assembling the parts after it has been fabricated. The chosen material was PLA as the design such as the gears needed the material to be strong and durable.

#### 4.2.6. CAD Design Dimensions

Figure 60 shows the completed CAD design of prototype 3. Table 8 below shows that the lift is more than the weight thus the prototype fabrication can proceed.



**Figure 60.** CAD design prototype 2.

**Table 8.** Lift and weight comparison.

Component	Weight (g)	Parameter	Value
Brushless Outrunner Motor	20	Flapping Amplitude (degree)	50
Radio Receiver	11.5	Flapping Frequency (Hz)	10
Servos	4.5	Lift Coefficient	0.8
Li-Po Battery	4	Air Density ( $\text{kg/m}^3$ )	1.225
Electronic Speed Controller	10	Chord Length (m)	0.1
MAV Design	14.77	WingSpan (m)	0.15
Total Weight	64.77	Lift (g)	977

#### 4.2.7. Flight Test

Figure 61 shows the assembled prototype. Similarly a dry run test was done for the flapping mechanism. Everything was working normally. Next it was the tethered flight. The MAV was also able to move in a circular motion. Finally the free flight test was carried out via remote control. The MAV was held until it flapped at high frequency after which it was hand thrown in the forward direction. After it was thrown, the MAV continued to fly forward while slowly pitching upwards.

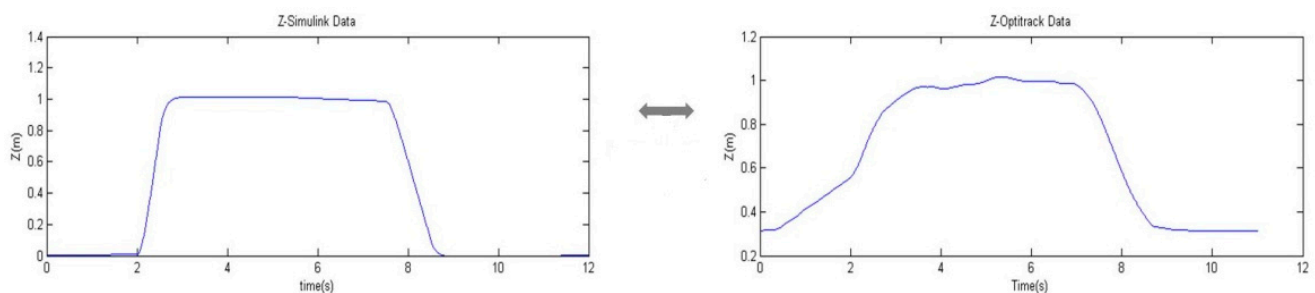
The left and right controls were tested and the MAV showed that it could maneuver left and right. The last test was the pitching control. Increasing the rpm of the motor pitches the MAV upwards and decreasing the rpm pitches the MAV downwards. The test was a success, the MAV showed that it can fly and was able to be controlled remotely. The video clip can be seen at Youtube website: <http://youtu.be/hp-Kpw6sll0>.



**Figure 61.** Assembled prototype 2.

Figure 62 shows the flight trajectory captured using Optitrack in our lab. The left figure shows our intended flight path. The right figure shows the actual trajectory of its body. The up/down motion due to the upstroke and downstroke of the wings at 10 Hz flapping frequency has been filtered out using notch filter, hence, the trajectory appears smooth as shown in the right figure. The gradual increase during  $1.0 < t \text{ (s)} < 2.0$  is due to initial throw to gain altitude and speed by the pilot. Subsequently,  $(2.5 < t \text{ (s)} < 7)$  the flapping wing flies at steady altitude and land ( $t > 7 \text{ s}$ ). It is clear that the second prototype was able to take off, climb, cruise and land in flapping mode successfully.

Besides unstable flight, certain segments of the flight test were captured well by the Optitrack system. It was then imported into MATLAB. In Figure 62 below the plots on the right segment was captured from three different test flights while the plots on left segment was obtained from the Simulation performing similar outcomes of the captured data.



**Figure 62.** Flight trajectory of the second flapping wing prototype using Optitrack. Left: Intended flight path simulation; Right: Actual flight path captured by Optitrack.

## 5. Conclusions

This paper reports the research and development of our in-house near-resonance type albatross-like flapping wing models for MAV. The flapping wing models mimic the long-distance migratory bird, similar to albatross. CFD results show that the albatross generates lift on its wing mainly by vortex lift mechanism. They do maneuvering by flapping its entire left and right wings at different amplitudes than using (flapping or twisting) its wing tip only. During forward motion, the wings produce a largely tilted leading edge vortex ring. The flight dynamic parameters is estimated, and used as guidance to predict flying characteristics of this type of ornithopter-like flapping wing MAV. With CFD results, we designed, built and flew two near resonance flapping wing MAVs. To test the flapping wing mechanism, a test cell was made to house the prototype and the load cell. When measuring the aerodynamic forces produced in the experiments, it was found that thrust was constantly generated, while lift was periodic in nature following a sinusoidal trend. It was found that lift is predominantly generated on the downstroke, with negative lift being generated on the upstroke. It was found out that the thin wing has both lift and thrust produced on than the PET film and thick cambered wing. Flexible wing generated higher velocities, frequency, lift and thrust. In observing the wing angle motion, it was found out that the lift occurs most when the wing is at  $0^\circ$  and  $-10^\circ$ , while negative lift at  $30^\circ$  and  $45^\circ$ .

The design sections of prototypes 2 and 3 have been discussed and evaluated the conceptual designs. There were two fabrication methods that were used, laser cutting and 3D printing. Although it seemed that the 3D printing was a better fabrication method as it allows for more complicated design it does has its limitations in the area of melting point and breaking strength. The third prototype could withstand the high frequency flapping and near resonance amplitude as designed. With remote control, the third prototype was able to take off, climb, cruise and land in flapping mode successfully.

## Author Contributions

In this paper, Woei-Leong Chan did the experiment on thrust measurement and material selection. Woei-Leong Chan also constructed the first flapping wing model. Sutthiphong Srigrarom made the second flapping wing model and wrote this paper. Notwithstanding, both authors worked together, therefore, the contribution and credit are equally shared.

## Conflicts of Interest

The authors declare no conflict of interest.

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